

SHROUD ASSEMBLY AND METHOD OF MACHINING SAME

ABSTRACT OF THE DISCLOSURE

A method of machining an inner surface (56) of a shroud assembly (42) extending generally circumferentially around a central axis (18) of a gas turbine aircraft engine (10). The method includes determining pre-machined radial clearances (60) during flight between tips (26) of rotor blades (20) in the engine (10) and the inner surface (56) of the shroud assembly (42) at each of a plurality of circumferentially spaced locations around the shroud assembly (42). The inner surface (56) of the shroud assembly (42) is machined based on the pre-machined radial clearances (60) to provide a generally uniform post-machined radial clearance (60) during flight between the tips (26) of the rotor blades (20) and the inner surface (56) of the shroud assembly (42) at each of the circumferentially spaced locations around the shroud assembly (42).

Figure 2