CLAIMS

What is claimed is:

1. A method to fill a gap in a component comprising:

vibrating the component; and

applying an alloy powder onto the component such that the alloy powder is received into a gap in response to the vibrating of the component.

2. The method as recited in claim 1, further comprising:

capping the alloy powder within the gap with a bi-modal braze composition.

3. The method as recited in claim 2, further comprising:

applying the bi-modal braze composition as a bead.

4. The method as recited in claim 3, further comprising:

applying the bi-modal braze composition with a syringe.

5. The method as recited in claim 2, further comprising:

utilizing a combination of an alloy powder; an alloy powder with a melting point depressant; and a braze binder as the bi-modal braze composition.

6. The method as recited in claim 2, further comprising:

utilizing a combination of 50-80% alloy powder; a 10% braze binder; and a remainder of an alloy powder with a melting point depressant as the bi-modal braze composition.

7. The method as recited in claim 2, further comprising:

orienting the component so the gap is generally upwards with respect to gravity.

8. The method as recited in claim 2, further comprising:

brazing the component within a furnace.

9. A method for repairing a gas turbine engine component comprising:

vibrating the aerospace component;

applying an alloy powder onto the component such that the alloy powder is received into a gap in response to the vibrating of the aerospace component; and

capping the alloy powder within the gap.

10. The method as recited in claim 9, further comprising:

orienting the component so the gap is generally upwards with respect to gravity.

11. The method as recited in claim 10, further comprising:

applying the bi-modal braze composition as a bead.

12. The method as recited in claim 11, further comprising:

applying the bi-modal braze composition with a syringe.

13. The method as recited in claim 10, further comprising:

brazing the component within a furnace.

14. The method as recited in claim 10, wherein the aerospace component is a vane.

15. The method as recited in claim 10, wherein the aerospace component is a blade.

16. A gas turbine engine component comprising:

a gap filled with a melted alloy powder and capped with a bi-modal braze composition.

17. The gas turbine engine component as recited in claim 16, wherein the aerospace component is a blade.

18. The gas turbine engine component as recited in claim 16, wherein the aerospace component is a vane.

CLAIMS

1. A method to fill a gap in a component comprising:

orienting the component so the gap is generally upwards with respect to gravity;

vibrating the component; and

applying an alloy powder onto the component such that the alloy powder is received into a gap in response to the vibrating of the component.

2. The method as recited in claim 1, further comprising:

capping the alloy powder within the gap with a bi-modal braze composition.

3. The method as recited in claim 2, further comprising:

applying the bi-modal braze composition as a bead.

4. The method as recited in claim 3, further comprising:

applying the bi-modal braze composition with a syringe.

5. The method as recited in claim 2, further comprising:

utilizing a combination of an alloy powder; an alloy powder with a melting point depressant; and a braze binder as the bi-modal braze composition.

6. The method as recited in claim 2, further comprising:

utilizing a combination of 50-80% alloy powder; a 10% braze binder; and a remainder of an alloy powder with a melting point depressant as the bi-modal braze composition.

7. The method as recited in claim 2, further comprising:

brazing the component within a furnace.

8. A method for repairing a gas turbine engine component comprising:

orienting the component so that a gap in the component is generally upwards with respect to gravity;

vibrating the component;

applying an alloy powder onto the component such that the alloy powder is received into the gap in response to the vibrating of the component; and

capping the alloy powder within the gap.

9. The method as recited in claim 8, further comprising:

applying a bi-modal braze composition as a bead.

10. The method as recited in claim 9, further comprising:

applying the bi-modal braze composition with a syringe.

11. The method as recited in claim 8, further comprising:

brazing the component within a furnace.

12. The method as recited in claim 8, wherein the component is a vane.

13. The method as recited in claim 8, wherein the component is a blade.