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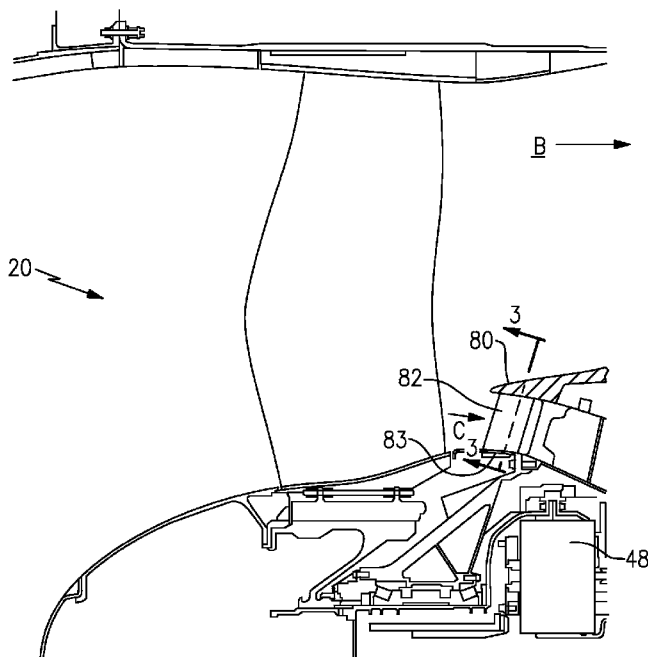
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(54) Title: ANTI-ICING CORE INLET STATOR ASSEMBLY FOR A GAS TURBINE ENGINE



(57) Abstract: A gas turbine engine is defined wherein the inlet guide vanes leading into a core engine flow path are sized and positioned such that flow paths positioned circumferentially intermediate the vane are sufficiently large that a hydraulic diameter of greater than or equal to about 1.5 is achieved. This will likely reduce the detrimental affect of icing.