



(43) International Publication Date
7 November 2013 (07.11.2013)

(51) International Patent Classification:

F02C 7/18 (2006.01) *F01D 25/12* (2006.01)
F01D 5/08 (2006.01) *F02K 3/00* (2006.01)

(21) International Application Number:

PCT/US2013/025876

(22) International Filing Date:

13 February 2013 (13.02.2013)

(25) Filing Language:

English

(26) Publication Language:

English

(30) Priority Data:

61/599,242 15 February 2012 (15.02.2012) US
13/544,153 9 July 2012 (09.07.2012) US

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(81) Designated States (unless otherwise indicated, for every kind of national protection available): AE, AG, AL, AM,

AO, AT, AU, AZ, BA, BB, BG, BH, BN, BR, BW, BY, BZ, CA, CH, CL, CN, CO, CR, CU, CZ, DE, DK, DM, DO, DZ, EC, EE, EG, ES, FI, GB, GD, GE, GH, GM, GT, HN, HR, HU, ID, IL, IN, IS, JP, KE, KG, KM, KN, KP, KR, KZ, LA, LC, LK, LR, LS, LT, LU, LY, MA, MD, ME, MG, MK, MN, MW, MX, MY, MZ, NA, NG, NI, NO, NZ, OM, PA, PE, PG, PH, PL, PT, QA, RO, RS, RU, RW, SC, SD, SE, SG, SK, SL, SM, ST, SV, SY, TH, TJ, TM, TN, TR, TT, TZ, UA, UG, US, UZ, VC, VN, ZA, ZM, ZW.

(84) Designated States (unless otherwise indicated, for every kind of regional protection available): ARIPO (BW, GH, GM, KE, LR, LS, MW, MZ, NA, RW, SD, SL, SZ, TZ, UG, ZM, ZW), Eurasian (AM, AZ, BY, KG, KZ, RU, TJ, TM), European (AL, AT, BE, BG, CH, CY, CZ, DE, DK, EE, ES, FI, FR, GB, GR, HR, HU, IE, IS, IT, LT, LU, LV, MC, MK, MT, NL, NO, PL, PT, RO, RS, SE, SI, SK, SM, TR), OAPI (BF, BJ, CF, CG, CI, CM, GA, GN, GQ, GW, ML, MR, NE, SN, TD, TG).

Published:

— without international search report and to be republished upon receipt of that report (Rule 48.2(g))

(54) Title: GAS TURBINE ENGINE COMPONENT WITH DIFFUSIVE COOLING HOLE

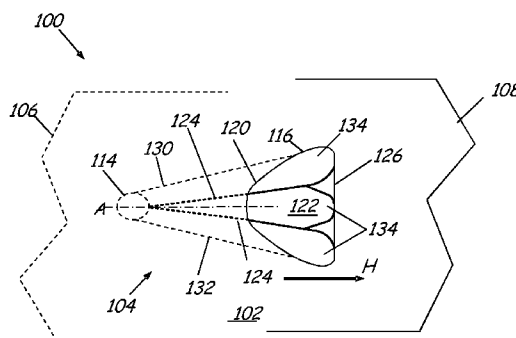


FIG. 4A

(57) Abstract: A component for a gas turbine engine has a cooling hole extending through a gas path wall. The gas path wall has a first surface and second surface, where the second surface is exposed to hot gas flow. The cooling hole includes a passage extending through the gas path wall from an inlet in the first surface to an outlet in the second surface, and a longitudinal divider separates the passage into lobes. The passage diverges through the gas path wall, such that cross-sectional area of the cooling hole increases continuously from the inlet through the passage to the outlet.