WHAT IS CLAIMED IS:

1. A method for repairing a turbine engine comprising a rotating stage of airfoils and an outer band surrounding the rotating stage of airfoils,

the rotating stage of airfoils including a plurality of airfoils defining an outer end, the method comprising:

determining a measure of a gap between the outer ends of the plurality of airfoils and the outer band is greater than a predetermined amount; and

depositing an abradable material on the outer band in situ to reduce the measure of the gap between the outer ends of the plurality of airfoils and the outer band,

wherein depositing the abradable material on the outer band in situ comprises depositing the abradable material using a spray module attached to at least one airfoil of the plurality of airfoils, and depositing the abradable material on the outer band in situ further comprises rotating the plurality of airfoils while depositing the abradable material on the outer band in situ.

2. The method of claim 1, wherein depositing the abradable material on the outer band in situ comprises depositing the abradable material using a spray module attached to at least one airfoil of the plurality of airfoils.

3. The method of claim 2, wherein depositing the abradable material on the outer band in situ further comprises rotating the plurality of airfoils while depositing the abradable material on the outer band in situ.

4. The method of claim 1, wherein the turbine engine defines a radial direction and a circumferential direction, and wherein determining the measure of the gap between the outer ends of the plurality of airfoils and the outer band comprises determining a two-dimensional measure of the gap along the radial direction and the circumferential direction.

5. The method of claim 1, wherein the turbine engine defines an axial direction, a radial direction, and a circumferential direction, and wherein determining the measure of the gap between the outer ends of the plurality of airfoils and the outer band comprises determining a three-dimensional measure of the gap along the axial direction, the radial direction, and the circumferential direction.

6. The method of claim 1, further comprising:

removing in situ at least a portion of the existing abradable material.

7. The method of claim 1, further comprising:

recontouring in situ the abradable material deposited on the outer band.

8. The method of claim 7, wherein recontouring in situ the abradable material deposited on the outer band comprises recontouring in situ the abradable material deposited on the outer band utilizing a recontouring tool attached to at least one airfoil of the plurality of airfoils.

9. The method of claim 8, wherein recontouring in situ the abradable material deposited on the outer band further comprises rotating the plurality of airfoils while recontouring in situ the abradable material deposited on the outer band.

10. The method of claim 1, wherein the plurality of airfoils of the rotating stage of airfoils is a plurality of rotor blades.

11. The method of claim 1, wherein depositing the abradable material on the outer band in situ comprises depositing the abradable material in situ using a snake arm tool extending through an opening of the turbine engine.

12. A method for repairing a turbine engine comprising a rotating stage of airfoils and an outer band surrounding the rotating stage of airfoils, the rotating stage of airfoils including a plurality of airfoils defining an outer end, the method comprising:

determining a measure of a qap between the outer ends of the plurality of airfoils and the outer band is greater than a predetermined amount; and

depositing an abradable material on the outer band in situ to reduce the measure of the gap between the outer ends of the plurality of airfoils and the outer band,

wherein the turbine engine defines a circumferential direction, and wherein depositing the abradable material on the outer band in situ comprises depositing the abradable material in a nonuniform manner along the circumferential direction on the outer band.

13. The method of claim 12, wherein depositing the abradable material in a nonuniform manner along the circumferential direction comprises depositing the abradable material in a nonuniform manner along the circumferential direction to result in a nonuniform measure of the gap along the circumferential direction.