CLAIMS:

1. A space vehicle comprising:

an optical system having a field of view, the optical system comprising at least two optical elements spaced from one another along an optical axis of the optical system, thereby defining an interior cavity of the optical system;

a plurality of holding assemblies, each of the plurality of holding assemblies comprising a folding mechanism configured and operable to move between a folded position corresponding to an inoperative mode of the optical system, and a deployed position corresponding to an operative mode of the optical system; and

a plurality of control systems each comprising at least one physical element configured for performing one or more functions for enabling operation of the space vehicle, each of the plurality of control systems configured to be deployable and mounted on a respective one of the plurality of holding assemblies operable to move between the folded position and the deployed position;

wherein, when in the folded position of the folding mechanism, at least one of the plurality of control systems being held by a respective one of the plurality of holding assemblies is at least partially located in said interior cavity of the optical system for stowage; and

wherein, when in the deployed position of the folding mechanism, said at least one of the plurality of control systems being held by the respective one of the plurality of holding assemblies is located outside said interior cavity and outside said field of view of the optical system, thereby allowing operation of the optical system.

2. The space vehicle of claim 1, wherein said optical system comprises a telescope having a primary optical element and a secondary element; said at least one of the plurality of control systems being positioned in between said primary and secondary elements.

3. The space vehicle of claim 1, wherein said plurality of control systems comprise at least one of a processing unit configured to control space vehicle operations, an image processing module configured to process imaging data collected by said optical system, a guiding module configured for guiding said vehicle, at least a receiving/transmitting device configured for receiving/transmitting signals, an amplifier

device operable for amplifying signals received by the receiver; and at least one power source configured for supplying voltage supply to the vehicle.

4. The space vehicle of claim 1, wherein said folding mechanism is configured and operable for being pivotally displaceable around an axis perpendicular to said optical axis.

5. The space vehicle of claim 1, wherein said folding mechanism is configured and operable for being pivotally displaceable around an axis parallel to said optical axis.

6. The space vehicle of claim 1, wherein said folding mechanism is configured and operable for being longitudinally displaceable along an axis perpendicular to said optical axis.

7. The space vehicle of claim 1, further comprising a frame comprising a plurality of surfaces forming an internal cavity and at least one opening in the frame, wherein said at least one of the plurality of holding assemblies is mounted on at least one of the plurality of surfaces and at least partially covers said at least one opening when said plurality of control systems are in its folded position.

8. The space vehicle of claim 7, wherein said folding mechanism of at least one of the plurality of holding assemblies comprises at least one support surface carrying a respective one said plurality of control systems and an axial hinge connecting between one of the surfaces of the frame and the support surface.

9. The space vehicle of claim 8, wherein in the deployed position said at least one support surface is substantially perpendicular to said optical axis of the optical system.

10. The space vehicle of claim 8, wherein in said deployed position said at least one support surface is substantially parallel to said optical axis of the optical system.

11. The space vehicle of claim 8, wherein said axial hinge has a main direction substantially parallel to said optical axis.

12. The space vehicle of claim 8, wherein said axial hinge has a direction substantially perpendicular to said optical axis.

13. The space vehicle of claim 1, wherein the space vehicle is configured as a satellite or a spacecraft.