

TURBINE COMPONENT WITH TIP FLAGGED PEDESTAL COOLING

ABSTRACT OF THE DISCLOSURE

A turbine blade for a gas turbine engine is provided with improved cooling at a radially outer portion of the trailing edge. A tip flag path is provided with a first pedestal array to increase the heat transfer at the area. In addition, a second pedestal array replaces the metering holes which has been positioned radially inwardly of the tip strip. The two pedestal arrays provide improved heat transfer at an area which has been subject to burning and spallation in the prior art.

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Figure 1