

WELDING REPAIR METHOD FOR FULL HOOP STRUCTURES

BACKGROUND OF THE INVENTION

This application relates to a method of repairing a full hoop structure with a
5 welding process, wherein heat treating is performed both at the location of the weld,
and at a diametrically opposed location.

Welding methods are sometimes necessary to repair metallic structures. As
an example, a cast part may have a defect such as shrinkage that may have occurred
in a cast mold. Alternatively, small cracks may form in the part.

10 Such operations are often used in components for gas turbine engines.
Structures that could be called “full hoop,” or structures that surround a central axis
for 360°, often require such welding procedures. Examples of such parts in a gas
turbine engine would be a diffuser case and a turbine exhaust case.

The weld being performed at a location on the part may cause an
15 unacceptably high residual stress. In the prior art, this stress has been relieved by
some post-weld heat treatment.

In one prior art method, the entire structure has been heated isothermally to
heat-treat temperatures. Heating isothermally does not induce additional thermal
stress at the weld, so the residual stress remains constant until actual heat treatment
20 takes place. This “global” heating can affect dimensions that have been “machined”
into the part by causing their residual stresses to also relax. In many cases, it has not
been found practical due to cost and complexity to fixture the part during heat
treatment to hold these dimensions constant.

Thus, localized heat treatment has also been utilized to avoid loss of
25 dimensions. Local heat treatment can have unforeseen and potentially detrimental
effects on the intended stress relaxation. The region being heated locally will
expand due to its temperature change. The surrounding non-heated material will
resist this expansion causing the heated area to become more compressively loaded.
Since the residual stress due to weld is tensile, the net effect of local heating is to
30 temporarily reduce the value of the tensile stress in the weld. If sufficient care is not
exercised, it is possible to reduce the value of the tensile stress so much so as to
eliminate it completely. In this case, subsequent heat treatment for stress relaxation

would be ineffective since the stress would already be reduced to zero. Note that the full value of the residual stress in this case would return when the locally applied temperature was removed.

Also of concern, would be a situation in which the weld stress was reduced
5 by local heating through zero and into a state of compression. This stress would relax during subsequent heat treatment, but this is far from the original intent of the heat treatment process, which was to reduce the tensile residual stress associated with the weld.

10 SUMMARY OF THE INVENTION

In the disclosed embodiment of this invention, a weld repair is made on a part with a full hoop structure. After the weld has been completed, heat-treating is performed at the location of the weld, and at the same time, at a second opposed location. In a disclosed embodiment, the second location is diametrically opposed to
15 the weld location. This heat-treating is preferably confined to as narrow a band as possible through the weld and its heat affected zone, and in a similar manner, at an opposed position to it. Furthermore, the heat-treating preferably occurs along an entire axial length of the part.

The opposed bands of heat-treating eliminate the compressive stresses
20 mentioned above from forming. This allows the modified local heat treatment to mimic the beneficial effect of a global heat treatment as mentioned above while avoiding the inherent problems.

While in the disclosed embodiment the part is a full hoop part, the present invention is more powerful, and extends beyond any particular shape of part. In
25 fact, an arbitrarily shaped part could benefit from this present invention. In an arbitrarily shaped part, an area of material on the part would be identified about which the part would thermally expand while not creating additional stress in the part at a weld treatment location. The weld treatment would be provided, and simultaneously, a local heat treatment would be provided at an area of the weld
30 treatment, and at the identified area.

In other optional embodiments, the second band could be a plurality of bands, which are displaced from the diametrically opposed location. As an example,

two separate bands spaced equally about a location spaced 180° from the weld treatment area could be utilized rather than a single second band.

In yet another embodiment, the second band can extend for a greater circumferential extent than the band about the weld treatment. In this manner, the heat treating on the second band can be at a lower temperature. By utilizing a lower temperature, the potential resultant dimensional changes in that second region can be reduced. Such dimensional changes are related to temperature, and thus being able to utilize a lower temperature, albeit over a larger area, might prove beneficial under certain applications.

These and other features of the present invention can be best understood from the following specification and drawings, the following of which is a brief description.

BRIEF DESCRIPTION OF THE DRAWINGS

Figure 1 schematically shows a gas turbine engine.

Figure 2 is a schematic view of a full hoop part.

Figure 3 is a cross-sectional view of a heat treatment occurring on the mentioned part.

Figure 4 shows yet another embodiment.

Figure 5 shows yet another embodiment.

DETAILED DESCRIPTION OF THE PREFERRED EMBODIMENT

A gas turbine engine 10 is illustrated in Figure 1 extending along an axial center line 12. A fan section 14 is upstream of a compressor section 16, a combustion section 18, and a turbine section 11. As is known, many components of a gas turbine engine 10 could be said to have a “full hoop” structure. The full hoop is defined as a structure that surrounds the axial center line 12 for 360°. An example of such full hoop structures found in the gas turbine engine 10 would include a diffuser case located downstream of the compressor, or a turbine exhaust case located downstream of the turbine section 11. The term “full hoop” should not be taken as requiring that the component would be cylindrical. In fact, the disclosed components could be better described as somewhat conical. Even that shape is not a

limitation on the definition of “full hoop” which could extend to non-symmetrical structures, or structures with complex surfaces and multi-faceted shapes at their outer surfaces.

As shown in Figure 2, such a part 50 can have defects such as a crack shown at 52. Other type defects may be a casting defect such as may be caused by shrinkage. A worker of ordinary skill in the art would recognize many of the known defects, which could require welding repair treatment.

As shown in Figure 3, a weld treatment 53 is being applied schematically by welding tool 60 at the crack 52. With the present invention, and after completion of the welding treatment, two narrow bands of heat treatment are applied at diametrically opposed locations 54 and 56. Preferably the circumferential extent of the bands is selected to only be wide enough to provide the stress relief at the weld joint 53 along the defect 52. Thus, the bands may well have the same circumferential width. As shown, heating structures 58 create these two heat treat locations. The heating structures may be induction coils, radiant lamps, gas burners, etc. The heat treatment can be on the order of 1500° F, although the heat treat temperatures may be as known in the art. The bands 54 and 56 extend along the entire length of the part 50, as shown in Figure 2. Of course, it may also be that the bands do not extend for the entire length of the part.

The present invention, by utilizing the two diametrically opposed bands, achieves the benefits provided by the global heating of the prior art, but also avoids the problems of global heating as encountered in the prior art.

Also, while the present invention is disclosed as being directed to full hoop parts, it would have benefits in certain parts that do not have the full hoop structure as defined above. Arbitrarily shaped parts could benefit from the present invention by heat treating two distinct zones, to allow the numerical value of weld residual stress to be heat treated, while greatly reducing or eliminating the liability of resultant dimensional changes. For non-full hoop structures, a line or plane of material to be locally heat treated as the second band, is the line or plane about which the structure would thermally expand without creating additional stress in the component at the weld. A worker of ordinary skill in this art can determine this line or plane with structural analysis.

Figure 4 shows another embodiment wherein the “second band” is actually provided by two separate bands 202 and 204. As can be appreciated, the two separate bands are disclosed as being spaced equally about the point P spaced 180° from the weld treatment area T. By positioning these separate bands about the point P, the beneficial effects provided by the above-disclosed embodiment can be achieved.

Figure 5 shows yet another embodiment wherein the circumferential extent of the second band 300 is wider than the circumferential extent of the weld band 302. The temperature provided at the second band 300 can be lower, such that potential resultant dimensional changes in this second band are reduced.

Again, a worker of ordinary skill in the art would recognize how to incorporate the optional embodiments of Figures 4 and 5 to best effect.

Although a preferred embodiment of this invention has been disclosed, a worker of ordinary skill in this art would recognize that certain modifications would come within the scope of this invention. For that reason, the following claims should be studied to determine the true scope and content of this invention.

CLAIMS

1. A method of treating a full hoop part comprising the steps of:
 - (1) identifying a defect in a part, said part extending for 360° about an
5 axial center line;
 - (2) performing a weld treatment on said defect to correct said defect; and
 - (3) performing heat treatment at a first band associated with said weld
treatment, and at a second band, said second band being selected to be spaced about
said axial center line from said first band.
- 10 2. The method as set forth in claim 1, wherein said second band includes a
location spaced by 180° about said axial center line from said first band.
3. The method as set forth in claim 1, wherein said first and second bands
15 receive the heat treatment concurrently.
4. The method as set forth in claim 1, wherein said second band has a greater
circumferential extent than said first band.
- 20 5. The method as set forth in claim 4, wherein said second band is heat treated
at a lower temperature than said first band.
6. The method as set forth in claim 1, wherein said first and second bands are
selected to be of roughly equal circumferential extent.
- 25 7. The method as set forth in claim 1, wherein said part extends for an axial
distance, and said first and second bands extending for the entirety of said axial
distance.
- 30 8. The method as set forth in claim 1, wherein a circumferential extent of said
first band is selected to only be wide enough to provide a stress relief in said weld
joint.

9. The method as set forth in claim 1, wherein said part is a gas turbine engine part.
10. The method as set forth in claim 5, wherein said part is a diffuser case.
- 5 11. The method as set forth in claim 5, wherein said part is a turbine exhaust case.
12. The method as set forth in claim 1, wherein heat treatment is also performed
- 10 at a third band spaced from both said first and second bands.
13. The method as set forth in claim 12, wherein said second and third bands are positioned to be equally spaced from said first band.
- 15 14. A component comprising:
a component body having an axial center line, and extending for 360° about said axial center line;
a weld treatment at a circumferential location within said 360°, said component body being heat-treated after formation of said weld treatment at a first
- 20 band incorporating said weld joint, and at a second band spaced from said first band about said axial center line.
15. The component as set forth in claim 14, wherein said first and second bands are selected to be of approximately equal circumferential extent.
- 25 16. The component as set forth in claim 14, wherein said component is a gas turbine engine part.
17. The component as set forth in claim 16, wherein said component is a diffuser
- 30 case.

18. The component as set forth in claim 16, wherein said component is a turbine exhaust case.

19. A method of providing a weld treatment on a part comprising the steps of:

- 5 (1) identifying a part in need of a weld treatment to correct a defect;
- (2) identifying an area of material on the part about which the part would thermally expand while not creating additional stress in the arbitrarily shaped part at the weld treatment;
- (3) providing the weld treatment; and
- 10 (4) simultaneously providing a local heat treatment at an area receiving the weld treatment, and at the identified area.

20. The method as set forth in claim 19, wherein the part is a full hoop part surrounding a central axis by 360°.

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