CLAIMS

1. A blade, comprising:

a blade body extending from a blade root to an opposed blade tip surface along a longitudinal axis, wherein the blade body defines a pressure side and a suction side, and wherein the blade body includes a cutting edge defined between the blade tip surface of the blade body and the pressure side of the blade body, wherein the cutting edge is configured to abrade a seal section of an engine case.

2. A blade as recited in claim 1, further comprising cutting points extending axially from the blade tip surface along the longitudinal axis.

3. A blade as recited in claim 2, further comprising a coating disposed on a portion of the blade tip surface, wherein the coating includes at least one of TiN, TiCN, TiAlN, Al_2O_3, CBN and diamond.

4. A blade as recited in claim 3, wherein the coating is disposed only on a portion of the blade tip surface that includes the cutting points.

5. A blade as recited in claim 1, wherein the blade tip surface includes a chamfered surface between the pressure side and the suction side of the blade body that tapers toward the blade root in a direction from the pressure side to the suction side.

6. A blade as recited in claim 5, wherein the blade tip surface includes a land on the blade tip surface between the pressure side and the chamfered surface.

7. A blade as recited in claim 6, wherein a portion of the land is at a ninety degree angle with respect to a portion of the pressure side of the blade body.

8. A blade as recited in claim 5, wherein the cutting edge defines an arcuate portion transitioning between the pressure side and a land of the blade tip surface, wherein the land is between the pressure side and the chamfered surface.

9. A blade as recited in claim 5, wherein cutting points extending axially from the blade tip surface along the longitudinal axis are disposed only on a land of the blade tip surface, wherein the land is on the blade tip surface between the pressure side and the chamfered surface.

10. A blade as recited in claim 1, wherein the cutting edge includes a projection portion, wherein the projection portion extends from the pressure side of the blade body.

11. A method for manufacturing a blade, the method comprising:

forming an airfoil with a root and an opposed tip surface along a longitudinal axis, wherein the airfoil defines a pressure side and a suction side; and

forming a cutting edge between the tip surface of the airfoil and the pressure side of the airfoil, wherein the cutting edge is configured to abrade a seal section of an engine case.

12. A method as recited in claim 11, wherein forming a cutting edge includes machining a chamfered surface on the tip surface between the pressure side and the suction side, wherein machining a chamfered surface includes tapering the chamfered surface toward the root in a direction from the pressure side to the suction side.

13. A method as recited in claim 11, wherein forming a cutting edge includes machining an arcuate portion between the pressure side and a land, wherein the land is surface on the tip surface between the pressure side and a chamfered surface, wherein the chamfered surface is on the tip surface between the pressure side and the suction side.

14. A method as recited in claim 11, wherein forming a cutting edge includes machining a projection portion extending from the pressure side.

15. A method as recited in claim 11, wherein forming a cutting edge includes forging a chamfered surface between the pressure side and the suction side, wherein forging a chamfered surface includes tapering the chamfered surface toward the root in a direction from the pressure side to the suction side.

16. A method as recited in claim 11, wherein forming a cutting edge includes forging an arcuate portion between the pressure side and a land, wherein the land is surface on the tip surface between the pressure side and a chamfered surface, wherein the chamfered surface is on the tip surface between the pressure side and the suction side.

17. A method as recited in claim 11, wherein forming a cutting edge includes forging a projection portion extending from the pressure side.

18. A method as recited in claim 11, further comprising forming cutting points in the tip surface, wherein the cutting points extend axially from the tip surface along the longitudinal axis.

19. A method as recited in claim 11, further comprising coating a portion of the tip surface with a coating material including at least one of TiN, TiCN, TiAlN, Al_2O_3, CBN and diamond.

20. A gas turbine engine comprising:

a case defining a centerline axis;

an abradable liner disposed radially inward from the case including a layer of rub material disposed on an inner diameter of the abradable liner;

a hub radially inward of the case and the abradable liner; and

a plurality of blade bodies extending radially outward from the hub for rotation about the centerline axis, wherein each blade body extends from a respective blade root to an opposed respective blade tip surface along a respective longitudinal axis, wherein each blade body defines a respective pressure side and a respective suction side, wherein each blade body includes a respective cutting edge defined between the blade tip surface and the pressure side of the blade body, wherein the cutting edge of each blade body is positioned proximate an inner diameter of the layer of rub material for abrading the layer of rub material during circumferential movement of the cutting edges as the blade bodies rotate about the centerline axis.