

AIRFOIL WITH LARGE FILLET AND MICRO-CIRCUIT COOLING

Abstract:

A gas turbine engine blade has a relatively large fillet to improve the characteristics of the air flow thereover. The fillet has a thin wall which partially defines a fillet cavity therebehind, and cooling air is provided to the fillet cavity and is then routed to the outer surface by way film cooling holes. Various design features are provided to enhance the effectiveness of the cooling air being provided to both the fillet cavity and other cavities within the blade.

Figures 4A & 4B