WHAT IS CLAIMED IS:

1. A system (100) for repairing an abradable material (70) coated on a casing (25, 40) of a gas turbine engine (10), the system (100) comprising:

an articulating guide (102) configured to fit into an access port (62) of the gas turbine engine (10), the articulating guide (102) comprising a proximal end (106) and a distal end (108); and

a repair tool (104) configured at the distal end (108) of the articulating guide (102), the repair tool (104) comprising a body (110) having a proximal end (112) and a shaped distal end (114), the shaped distal end (114) extending away from the body (110) and shaped to trench out an area of the abradable material (70) comprising a defect (72), wherein the body (110) of the repair tool (104) comprises a wire; and

a filler material (120) for filling the trenched out area (125);

characterised in that the system (100) further comprises

a heating element (122) configured to heat the wire for heating the abradable material (70) as the shaped distal end (114) trenches out the area of the abradable material (70) comprising the defect (62).

2. The system (100) of claim 1, wherein the shaped distal end (114) comprises a varying cross-section having a first portion and second portion, wherein the first portion comprises a larger cross-sectional area than the second portion.

3. The system (100) of claim 1 or claim 2, wherein the shaped distal end (114) comprises at least one of an I-shaped cross-section, a T-shaped cross-section, a V-shaped cross-section, or an L-shaped cross-section.

4. The system (100) of any preceding claim, wherein the shaped distal end (114) comprises sharpened edges configured to cut the abradable material (70).

5. The system (100) of any preceding claim, further comprising a vibration element configured to vibrate the wire of the repair tool (104).

6. The system (100) of any preceding claim, wherein the casing comprises at least one of a fan casing (40) or a compressor casing (25) of the gas turbine engine (10).

7. A method (200) for repairing an abradable material (70) coated on a casing (25, 40) of a gas turbine engine (10), the method (200) comprising:

inserting an articulating guide (102) into an access port (62) of the gas turbine engine (10), the articulating guide (102) having a repair tool (104) at a distal end thereof;

positioning the repair tool (104) near a defect (72) of the abradable material (70), the repair tool (104) having a body (110) with a proximal end and a shaped distal end (114), the shaped distal end (114) extending away from the body (110), wherein the body (110) of the repair tool (104) comprises a wire;

plunging the shaped distal end (114) into the abradable material (70) at the defect (62) so as to trench out the defect (72), thereby forming a trenched out area (125); and filling the trenched out area (125) with a filler material (120);

characterised in that the method further comprises:

heating, via a heating element (122), the wire of the repair tool (104); and

heating, via the heated wire, the abradable material (70) as the shaped distal end (114) is plunged into the abradable material (70).

8. The method (200) of claim 7, further comprising after plunging the shaped distal end (114) into the abradable material (70), dragging the repair tool (104) through the abradable material (70).

9. The method (200) of claim 7 or claim 8, further comprising after plunging the shaped distal end (114) into the abradable material (70), rotating the repair tool (104) in place in the abradable material (70).

10. The method (200) of any of claims 7 to 9, wherein the shaped distal end (114) comprises at least one of an I-shaped cross-section, a T-shaped cross-section, or an L-shaped cross-section.

11. The method (200) of any of claims 7 to 10, wherein the shaped distal end (114) comprises sharpened edges configured to cut the abradable material (70) when plunging the shaped distal end (114) into the abradable material (70).