

Abstract

The invention relates to a turbomachine blade (10), in particular a turbine blade of a gas turbine. The turbomachine blade (10) designed according to the invention comprises a blade root (11) with a cooling-fluid feed (12), a blade tip (13) which is formed with a shroud element (14). At least one throughflow passage (19a) which extends in the blade longitudinal direction is arranged inside the airfoil (15). A shroud cooling passage (22) is formed in the shroud element at least along a section of the latter, this shroud cooling passage (22) being connected via an opening (23, 24) to the throughflow passage (19a), and a cooling fluid flowing through this shroud cooling passage (22) during operation of the turbomachine blade. At least one flow-guiding element (30, 31-d1 - 31-d11, 31-s1 - 31-s12, 32) for guiding at least some of the cooling-fluid flow flowing through the shroud cooling passage is arranged in the shroud cooling passage and/or in an inflow or outflow to or from the shroud cooling passage.

(Figure 4)