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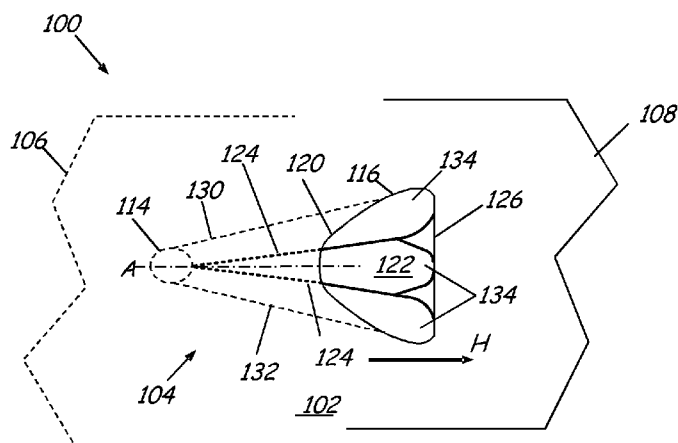


FIG. 4A

(57) Abstract: A component for a gas turbine engine has a cooling hole extending through a gas path wall. The gas path wall has a first surface and second surface, where the second surface is exposed to hot gas flow. The cooling hole includes a passage extending through the gas path wall from an inlet in the first surface to an outlet in the second surface, and a longitudinal divider separates the passage into lobes. The passage diverges through the gas path wall, such that cross-sectional area of the cooling hole increases continuously from the inlet through the passage to the outlet.