

Description

This invention relates to a rotor arrangement for a rotorcraft and in particular to an arrangement which comprises a plurality of intermeshing rotors having single lift surface blades.

The invention has broad application to various types of rotorcraft, including conventionally powered free flying helicopters, although it has been developed primarily for use in a craft which is intended to hover at the end of a tether. In the latter case the craft may be held aloft by the reaction between the rotors and high velocity winds or by motor drive applied to the rotors. Motor drive may be derived from an internal combustion engine or from one or more electric motors and, in the case of a tethered craft, energy may be supplied to the engine or motor by way of the tether. When an electric motor is employed to drive the craft aloft and the craft is placed in a suitable wind stream, the motor may be operated in a generating mode and be used to harness wind energy acting on the rotors. As an alternative, the craft, whether free flying or tethered, may be used simply to carry a sensor, such as a remotely controlled camera, or other device and, thus, the craft may be constructed in a size and manner appropriate to an unmanned craft.

Therefore, it is to be understood that the invention has broad application and need not be limited to use in conventional helicopters.

Various types of multi-rotor craft are known, including one which incorporates two intermeshing, two-bladed counter-rotating rotors with a 90° phase difference between the rotors. Also, craft have been developed which use rotors having blades which have a single lifting surface, including one craft which uses two spaced-apart such rotors with separate drive units for each of the rotors. Reference may be made to the publication "Vertical Flight", edited by Boyne, W.J. and Lopez, D.S. and published by the Smithsonian Institution Press, 1984 for particulars of these and other rotorcraft.

Also, reference may be made to US-A-3,053,480 which discloses a helicopter drone which has four intermeshing two-bladed rotors which are driven such that one diagonally disposed pair of the rotors co-rotate in a first direction whilst the other diagonally disposed pair of rotors co-rotate in the opposite direction.

In contrast with these known craft, the present invention is directed to one which employs four intermeshing rotors, with each rotor having a single blade, or, alternatively expressed, having a blade with a single lifting surface. Also, the rotors are mounted to respective rotor shafts which are divergently inclined with respect to one another in the

direction of the rotors. This arrangement provides a compact drive to all rotors from a single power source or from a number of coupled power sources.

Thus, the present invention may be defined broadly as providing a rotor craft comprising four intermeshing rotors coupled to common drive unit, each rotor being connected to an associated rotor shaft by way of a mechanism which provides for torque transmission and which permits rotor flapping and pitch adjustment. One pair of diagonally positioned rotors is arranged to be driven so as to co-rotate in one direction and the other pair of diagonally positioned rotors is arranged to be driven so as to co-rotate in the opposite direction, and means are provided for adjusting the pitch angle of each rotor. The rotorcraft is characterized in that the rotors are coupled to the driven unit by respective said rotor shafts, in that each rotor comprises a single rotatable blade having a lifting surface and an oppositely projecting substantially non-lifting balance arm, and in that the rotor shafts are inclined with respect to one another in a manner such that the rotor shafts diverge in the direction of the rotors whereby the path surface traced by each blade during rotation is inclined with respect to each other path surface.

The four-rotor arrangement of the present invention facilitates the employment of a relatively simple control system for the rotors, and the arrangement obviates the need for cyclic pitch control and complex mechanical systems which normally are required to effect cyclic pitch control. It is envisaged that collective pitch control only will be applied to the rotors, with differential such control being applied to various ones of the rotors in order to effect manoeuvring of the craft.

The invention will be more fully understood from the following description of an electrically powered rotorcraft of a type which is intended to carry a video camera for use in aerial photography or the like. The rotorcraft is illustrated schematically in the accompanying drawings in which:

Figure 1 shows a front elevation view of the craft,

Figure 2 shows a plan view of the craft as seen from above,

Figures 3 and 4 illustrate plan and side elevation views of a rotor which has been removed from an associated rotor shaft,

Figure 5 shows an elevation view of an upper portion of a drive shaft and a universal joint mounted to the drive shaft,

Figure 6 shows a plan view of the arrangement shown in Figure 5,

Figure 7 shows a mechanism which may be employed for effecting pitch control over the four rotors shown in Figure 2.

Figure 8 shows a schematic representation of an (alternative) electronic control system for effecting pitch control over the four rotors,

Figure 9 shows an arrangement for transferring drive to the rotor drive shafts from a single drive unit, and

Figures 10A and 10B show alternative methods of transmitting the drive to the drive shafts from the drive unit.

As shown in Figures 1 and 2 of the drawings, the rotorcraft comprises a body portion 20 which is mounted upon pods or skids 21. The body portion houses a drive motor and associated equipment, to be referred to in more detail below, and it is intended to house a camera (not shown) for use in aerial photography or reconnaissance. The craft incorporates four intermeshing rotors 22 to 25 which are rotatable about inclined axes 22A to 25A respectively. The two rotors 22 and 24 lie parallel to one another and the other two rotors 23 and 25 lie parallel to one another. Thus, the rotors 22, 24 and 23, 25 that have diagonally positioned axes of rotation are disposed parallel to each other, and at four instants in time during each revolution the adjacent rotors will be disposed at rightangles to one another, as shown in Figure 2.

The dotted outline in Figure 2 indicates the total area which is swept by the four rotors, and the dotted outline shown in Figure 1 indicates the maximum flapping angle through which each rotor blade travels.

The arrow which is shown in Figure 2 indicates the direction of intended forward travel of the craft. This direction is relevant in the context of the mechanisms shown in Figures 7 and 10 of the drawings and the arrows which are shown in the respective figures should all be taken as indicating one and the same direction of forward travel.

The four rotors 22 to 25 are coupled to a common drive unit 26 which is shown in Figure 9, and each rotor comprises a single rotating blade 27 having an oppositely projecting (integrally formed) radial balance arm 28. The blade 27 is shaped with a single lifting surface to provide aerodynamic lift. The radial balance arm 28 does not have a lifting surface and it serves solely to carry a balance weight 29 which functions to balance centrifugal forces which exist in rotation of the rotor. The balance arm 28 preferably has a length in the order of 20% to 35% of the length of the blade 27 and it is expected that the length of the balance arm will never exceed 40% of the length of the blade.

Each of the rotors 22, 25 extends through and is clamped in a universal joint 30, which will be described below with reference to Figures 5 and 6, and the rotor is formed with an aperture 31 which is centred on the axis of rotation of the rotor. The aperture 31 is intended to provide clearance ar-

ound a pitch control shaft 32 which projects through the aperture.

The rotors 22 to 25 are coupled to the drive unit 26 by respective drive shafts (i.e., rotor shafts) 33 to 36 and the shafts are inclined with respect to one another such that the path surface traced by each of the blades 27 is inclined with respect to each other path surface. Each of the shafts 33 to 36 is formed along at least a portion of its length as a tube so that it may accommodate the pitch control shaft 32, which extends longitudinally within the rotor shaft.

Various methods may be employed for imparting drive to the rotor shafts 33 to 36 from the drive unit 26, and one such method is shown schematically in Figures 9 and 10 of the drawings. Drive from the single drive unit 26 is imparted to all four of the rotor shafts by way of bevel gears 37. Two alternative approaches may be taken to the transmission of torque from the drive unit to the four rotor shafts 33 to 36, the preferred method being shown in Figure 10A and a second method being shown in Figure 10B. In the arrangement illustrated in Figure 10A two separate points of intersection occur for the four shafts 33 to 36, whereas in the arrangement shown in Figure 10B a single point of intersection exists for all four shafts.

Drive is transmitted to the shafts in such a manner that shafts 33 and 35 rotate in one direction and shafts 34 and 36 rotate in the opposite direction. That is, the adjacent shafts counter-rotate and the diagonally disposed shafts co-rotate.

The universal joint 30 is fixed to the upper end of each of the rotor shafts 33 to 36 and, as shown in Figures 5 and 6, the universal joint has a yoke 38 which provides a trunnion mounting for a blade carrier 39. The blade carrier pivots about the yoke axis 40 and, thus, permits see-sawing pivotal (flapping) movement of the blade 27 and its oppositely projecting radial arm 28.

The blade 27 and radial arm 28 are connected to the blade carrier 39 by clamp members 41 and 42, and such clamp members are pivotably connected to the blade carrier 39 by way of a pivot pin 43. This pivotal mounting allows the rotor blade 27 and the oppositely extending radial arm 28 to pivot about the longitudinal axis 44 of the rotor blade and, therefore, to adjust the pitch angle of the blade.

The clamp members 41 and 42 have laterally projecting lever arms 45 and the lever arms are connected to a terminal end 46 of the pitch control shaft 32 by a connecting elements 47. Rectilinear movement of the shaft 32 translates to pivotal movement of the lever arm 45 and, thus, upward and downward movement of the shaft 32 causes angular adjustment of the blade pitch.

Various different mechanisms may be em-

employed for imparting motion to the pitch control shaft 32 and one such mechanism is shown schematically in Figure 7 of the drawings.

In the arrangement shown in Figure 7, two of the blade pitch control shafts 32A and 32B are positively but rotatably connected by way of bearings 63 to a tilting plate 50 at points indicated by letters "A" and "B". The remaining two pitch control shafts 32C and 32D are rotatably connected to bell-crank levers 51 by way of bearings 64, and they meet (but are not directly connected to) the tilting plate at points "C", and "D".

The bell-crank levers 51 are pivotably connected to the tilting plate 50 by way of mounting arms 52, and the bell-crank levers 51 are connected to actuators 53. By operating the actuators 53, the pitch control shafts 32C and 32D are caused to move upwardly and downwardly with respect to the tilting plate 50.

The tilting plate 50 is itself supported by three further actuators 54, 55 and 56 which are located at the apices of a triangle. One of the arms 56 connects with a central point in the plate and the other two actuators 54 and 55 connect with the plate at peripheral points.

Uniform (simultaneous) actuation of all three of the actuators 54, 55 and 56 will result in collective pitch being applied to (or taken from) all four rotor blades. Also, differential collective pitch may be applied to the rotors by actuating any one or two of the three actuators 54 to 56. Thus, roll control of the craft may be exerted by the actuator 55, and pitch control of the craft may be exerted by actuator 54. Yaw control may be exercised by the actuators 53 and, of course, simultaneous actuation of a number of the actuators 53 through to 56 will result in compound movement of the craft.

As an alternative to the above described arrangements, the pitch control shafts 32 may be actuated by coupling each of the shafts to an associated actuator 60, as shown in Figure 8. Operation of the actuators 60 is in this case controlled by a microprocessor 61 to which inputs are provided from a controller 62. The controller 62 may derive its inputs from pilot actuated levers and/or pedals or from a remote transmitter.

The rotorcraft in accordance with the invention permits enhanced stability and controllability relative to the more conventional forms of rotorcraft. Thus, it can be shown for hovering operation that stability can be maintained in a four-rotor craft if the following relationship is established:

$$h/l \cos \epsilon = [(Zw/Xu)-1]/[(Zw/Xu) + \cot^2 \epsilon + \operatorname{cosec}^2 \epsilon]$$

where:

h = vertical height of rotor hubs above

centre of gravity of the craft.

l = virtual length of rotor shafts.

ϵ = angle of inclination (with respect to vertical axis) of the rotor shafts.

Xu = in-situ drag damping derivative of the rotors with respect to the axis of no-feathering.

Zw = in-situ vertical damping derivative of the rotors with respect to the axis of no-feathering.

Claims

1. A rotorcraft comprising four intermeshing rotors (22-25) coupled to common drive unit (26), each rotor (22-25) being connected to an associated rotor shaft (33-36) by way of a mechanism which provides for torque transmission and which permits rotor flapping and pitch adjustment, one pair of diagonally positioned rotors (22 and 24) being arranged to be driven so as to co-rotate in one direction and the other pair of diagonally positioned rotors (23 and 25) being arranged to be driven so as to co-rotate in the opposite direction, and means (32) being provided for adjusting the pitch angle of each rotor; characterized in that the rotors (22-25) are coupled to the driven unit by respective said rotor shafts (33-36), in that each rotor (22-25) comprises a single rotatable blade (27) having a lifting surface and an oppositely projecting substantially non-lifting balance arm (28), and in that the rotor shafts (33-36) are inclined with respect to one another in a manner such that the rotor shafts diverge in the direction of the rotors (22-25) whereby the path surface traced by each blade (27) during rotation is inclined with respect to each other path surface.
2. The rotorcraft as claimed in Claim 1 further characterized in that the rotor shafts (33-36) are centred geometrically on the corners of a square.
3. The rotorcraft as claimed in Claim 1 or Claim 2 further characterized in that the rotor shafts (33-36) are disposed symmetrically at each side of the fore-aft centre line of the rotor craft.
4. The rotorcraft as claimed in any one of claims 1 to 3 further characterised in that the rotors (22, 24 and 23, 25) which have diagonally positioned axes of rotation are, as viewed in plan, disposed parallel to each other.
5. The rotorcraft as claimed in any one of the preceding claims further characterised in that



employed for imparting motion to the pitch control shaft 32 and one such mechanism is shown schematically in Figure 7 of the drawings.

In the arrangement shown in Figure 7, two of the blade pitch control shafts 32A and 32B are positively but rotatably connected by way of bearings 63 to a tilting plate 50 at points indicated by letters "A" and "B". The remaining two pitch control shafts 32C and 32D are rotatably connected to bell-crank levers 51 by way of bearings 64, and they meet (but are not directly connected to) the tilting plate at points "C", and "D".

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Uniform (simultaneous) actuation of all three of the actuators 54, 55 and 56 will result in collective pitch being applied to (or taken from) all four rotor blades. Also, differential collective pitch may be applied to the rotors by actuating any one or two of the three actuators 54 to 56. Thus, roll control of the craft may be exerted by the actuator 55, and pitch control of the craft may be exerted by actuator 54. Yaw control may be exercised by the actuators 53 and, of course, simultaneous actuation of a number of the actuators 53 through to 56 will result in compound movement of the craft.

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The rotorcraft in accordance with the invention permits enhanced stability and controllability relative to the more conventional forms of rotorcraft. Thus, it can be shown for hovering operation that stability can be maintained in a four-rotor craft if the following relationship is established:

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where:

h = vertical height of rotor hubs above

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Claims

1. A rotorcraft comprising four intermeshing rotors (22-25) coupled to common drive unit (26), each rotor (22-25) being connected to an associated rotor shaft (33-36) by way of a mechanism which provides for torque transmission and which permits rotor flapping and pitch adjustment, one pair of diagonally positioned rotors (22 and 24) being arranged to be driven so as to co-rotate in one direction and the other pair of diagonally positioned rotors (23 and 25) being arranged to be driven so as to co-rotate in the opposite direction, and means (32) being provided for adjusting the pitch angle of each rotor; characterized in that the rotors (22-25) are coupled to the driven unit by respective said rotor shafts (33-36), in that each rotor (22-25) comprises a single rotatable blade (27) having a lifting surface and an oppositely projecting substantially non-lifting balance arm (28), and in that the rotor shafts (33-36) are inclined with respect to one another in a manner such that the rotor shafts diverge in the direction of the rotors (22-25) whereby the path surface traced by each blade (27) during rotation is inclined with respect to each other path surface.
2. The rotorcraft as claimed in Claim 1 further characterized in that the rotor shafts (33-36) are centred geometrically on the corners of a square.
3. The rotorcraft as claimed in Claim 1 or Claim 2 further characterized in that the rotor shafts (33-36) are disposed symmetrically at each side of the fore-aft centre line of the rotor craft.
4. The rotorcraft as claimed in any one of claims 1 to 3 further characterised in that the rotors (22, 24 and 23, 25) which have diagonally positioned axes of rotation are, as viewed in plan, disposed parallel to each other.
5. The rotorcraft as claimed in any one of the preceding claims further characterised in that



the balance arm (28) of each rotor (22-25) has a length not greater than 40% of the length of the oppositely projecting blade (27) and in that the balance arm (28) is fitted with a balance weight (29) which balances the centrifugal force of the blade during rotation of the rotor.

6. The rotorcraft as claimed in any one of the preceding claims further characterised in that the balance arm (28) and blade portions (27) of each rotor (22-25) are formed as an integral structure, in that an aperture (31) is formed within the rotor in alignment with the axis of rotation of the rotor, and in that the means (32) for adjusting the pitch angle of each blade projects through the aperture (31).

7. The rotorcraft as claimed in any one of the preceding claims further characterised in that the mechanism (38, 39) which connects each rotor (22-25) to its associated rotor shaft (33-36) comprises a universal joint, the universal joint having a yoke (38) connected to the rotor shaft, a rotor carrier (39) which is pivotably mounted to the yoke (38) in a manner which permits blade flapping motion and clamp members (41 and 42) securing the rotor (22-25) to the rotor carrier (39), the clamp members (41 and 42) being pivotably mounted to the rotor carrier (39) in a manner which permits adjustment of the blade pitch angle.

8. The rotorcraft as claimed in claim 7 further characterised in that the clamp members (41 and 42) have projecting lever arms (45) which are connected with the means (32) for adjusting the pitch angle of each blade, whereby rectilinear motion of said means (32) translates to angular adjustment of the blade pitch.

9. The rotorcraft as claimed in any one of the preceding claims further characterised in that the means for adjusting the pitch angle of the blades include pitch control shafts (32) which are located one within each of the rotor shafts (33-36), in that each pitch control shaft (32) is moveable in a rectilinear direction to transfer pitch adjustment motion to the associated blade and in that drive mechanisms (53-56, or 60) are coupled to the pitch control shafts (32) for imparting differential collective pitch adjustment motion to each of the pitch control shafts.

Revendications

1. Appareil volant à rotors comprenant quatre rotors imbriqués (22 à 25) accouplés à un dispositif moteur commun (26), chaque rotor (22 à

25) étant relié à un arbre de rotor associé (33 à 36) par l'intermédiaire d'un mécanisme qui assure une transmission de couple et permet le battement et le réglage de l'inclinaison du rotor, une paire de rotors placés diagonalement (22 et 24) étant arrangée pour être mue de façon à tourner ensemble dans un sens et l'autre paire de rotors placés diagonalement (23 et 25) étant arrangée pour être mue de façon à tourner ensemble dans l'autre sens, et un moyen (32) étant prévu pour le réglage de l'angle d'inclinaison de chaque rotor, caractérisé par le fait que les rotors (22 à 25) sont accouplés au dispositif moteur par les arbres de rotor respectifs (33 à 36), que chaque rotor (22 à 25) comprend une seule pale rotative (27) ayant une surface sustentatrice et un bras équilibreur opposé pratiquement non sustentateur (28), et que les arbres de rotor (33 à 36) sont inclinés les uns par rapport aux autres de façon à diverger en direction des rotors (22 à 25) afin que la surface balayée par chaque pale (27) pendant sa rotation soit inclinée par rapport à la surface balayée par chacune des autres pales.

2. Appareil volant à rotors selon la revendication 1, caractérisé en outre par le fait que les arbres de rotor (33 à 36) sont centrés géométriquement aux sommets d'un carré.

3. Appareil volant à rotors selon l'une des revendications 1 et 2, caractérisé par le fait que les arbres de rotor (33 à 36) sont disposés symétriquement de chaque côté de l'axe longitudinal de l'appareil.

4. Appareil volant à rotors selon l'une des revendications 1 à 3, caractérisé en outre par le fait que les rotors (22, 24 et 23, 25) qui ont des axes de rotation placés diagonalement sont, vus de dessus, disposés parallèlement l'un à l'autre.

5. Appareil volant à rotors selon l'une des revendications précédentes, caractérisé en outre par le fait que le bras équilibreur (28) de chaque rotor (22 à 25) a une longueur ne dépassant pas 40 % de la longueur de la pale opposée (27) et est pourvu d'un contrepoids (29) qui équilibre la force centrifuge de la pale pendant la rotation du rotor.

6. Appareil volant à rotors selon l'une des revendications précédentes, caractérisé en outre par le fait que le bras équilibreur (28) et des parties de pale (27) de chaque rotor (22 à 25) forment un seul bloc, qu'une ouverture (31) est



faite dans le rotor dans l'alignement de l'axe de rotation de celui-ci, et que le moyen (32) de réglage de l'angle d'inclinaison de chaque pale fait saillie dans cette ouverture (31).

7. Appareil volant à rotors selon l'une des revendications précédentes, caractérisé en outre par le fait que le mécanisme (38, 39) qui relie chaque rotor (22 à 25) à son arbre de rotor associé (33 à 36) comprend un joint universel, celui-ci ayant une chape (38) reliée à l'arbre de rotor, un support de rotor (39) qui est monté pivotant sur la chape (38) de manière à permettre le battement de la pale et des éléments de serrage (41 et 42) fixant le rotor (22 à 25) au support de rotor (39), ces éléments de serrage (41 et 42) étant montés pivotants sur le support de rotor (39) de manière à permettre le réglage de l'angle d'inclinaison de la pale.

8. Appareil volant à rotors selon la revendication 7, caractérisé en outre par le fait que les éléments de serrage (41 et 42) ont des bras de levier saillants (45) qui sont reliés au moyen (32) de réglage de l'angle d'inclinaison de chaque pale, de sorte qu'un mouvement rectiligne de ce moyen (32) se traduit par un réglage de l'angle d'inclinaison de la pale.

9. Appareil volant à rotors selon l'une des revendications précédentes, caractérisé en outre par le fait que les moyens de réglage de l'angle d'inclinaison des pales comportent des arbres de commande d'inclinaison (32) placés un dans chacun des arbres de rotor (33 à 36), que chaque arbre de commande d'inclinaison (32) est mobile en ligne droite pour transmettre le mouvement de réglage d'inclinaison à la pale associée, et que des mécanismes de commande (53 à 56 ou 60) sont accouplés aux arbres de commande d'inclinaison (32) pour communiquer à chacun de ceux-ci un mouvement de réglage d'inclinaison collectif différentiel.

Patentansprüche

1. Drehflügel-Flugzeug mit vier ineinandergreifend umlaufenden mit einer gemeinsamen Antriebseinheit (26) gekoppelten Rotoren (22-25), wobei jeder Rotor (22-25) mit einer zugehörigen Rotorwelle (33-36) über einen Mechanismus verbunden ist, der für Drehmomentübertragung sorgt und Einstellung des Schlagens und des Anstellwinkels für den Rotor zuläßt, wobei ein Paar diagonal angeordnete Rotoren (22 und 24) antriebsmäßig so ausgelegt ist,

daß sie in einer Richtung miteinander rotieren, und daß andere Paar diagonal angeordnete Rotoren (23 und 25) antriebsmäßig so ausgelegt ist, daß sie in der entgegengesetzten Richtung miteinander rotieren, und Mittel (32) vorgesehen sind zum Einstellen des Anstellwinkels jedes Rotors; dadurch gekennzeichnet, daß die Rotoren (22-25) mit der angetriebenen Einheit durch jeweilige Rotorwellen (33-36) gekoppelt sind, daß jeder Rotor (22-25) ein einziges drehbares Blatt (27) mit einer Auftriebsfläche und einen in entgegengesetzter Richtung abstehenden im wesentlichen nicht zum Auftrieb beitragenden Ausgleichsarm (28) besitzt, und daß die Rotorwellen (33-36) gegeneinander in einer solchen Weise geneigt sind, daß die Rotorwellen in Richtung der Rotoren (22-25) divergieren, wodurch die durch jedes Blatt (27) während der Drehung beschriebene Wegfläche gegen jede andere derartige Wegfläche geneigt ist.

2. Drehflügel-Flugzeug nach Anspruch 1, weiter dadurch gekennzeichnet, daß die Rotorwellen (33-36) geometrisch an den Ecken eines Quadrats zentriert sind.

3. Drehflügel-Flugzeug nach Anspruch 1 oder 2, weiter dadurch gekennzeichnet, daß die Rotorwellen (33-36) symmetrisch zu beiden Seiten der Mittellinie vorne/hinten des Drehflügel-Flugzeugs angeordnet sind.

4. Drehflügel-Flugzeug nach einem der Ansprüche 1 bis 3, weiter dadurch gekennzeichnet, daß die Rotoren (22-24 und 23-25), die jeweils diagonal angeordnete Drehachsen besitzen, in Draufsicht gesehen parallel zueinander angeordnet sind.

5. Drehflügel-Flugzeug nach einem der vorangehenden Ansprüche, weiter dadurch gekennzeichnet, daß der Ausgleichsarm (28) jedes Rotors (22-25) eine Länge von nicht mehr als 40% der Länge des entgegengesetzt abstehenden Blattes (27) besitzt und daß der Ausgleichsarm (28) mit einem Ausgleichsgewicht (29) versehen ist, das die Zentrifugalkraft des Blattes während der Drehung des Rotors ausgleicht.

6. Drehflügel-Flugzeug nach einem der vorangehenden Ansprüche, weiter dadurch gekennzeichnet, daß die Ausgleichsarm-(28) und Blatt-Abschnitte (27) jedes Rotors (22-25) als ein integraler Aufbau ausgebildet sind, und daß ein Durchbruch (31) innerhalb des Rotors in Ausrichtung mit der Drehachse des Rotors

ausgebildet ist, und daß das Mittel (32) zum Einstellen des Anstiegswinkels jedes Blattes durch den Durchbruch (31) vorsteht.

7. Drehflügel-Flugzeug nach einem der vorangehenden Ansprüche, weiter dadurch gekennzeichnet, daß der jeden Rotor (22-25) mit seiner zugeordneten Rotorwelle (33-36) verbindende Mechanismus (38, 39) ein Universalgelenk umfaßt, welches Universalgelenk aufweist:
 - ein mit der Rotorwelle verbundenes Joch (38),
 - einen Rotorträger (39), der an dem Joch (38) in einer die die Schlagbewegung des Blattes zulassenden Weise schwenkbar angebracht ist, und Klemmelemente (41 und 42), welche den Rotor (22-25) an dem Rotorträger (39) sicher befestigen, wobei die Klemmelemente (41 und 42) schwenkbar an dem Rotorträger (39) in einer Weise angebracht sind, die eine Einstellung des Blattansellwinkels erlaubt.

8. Drehflügel-Flugzeug nach Anspruch 7, weiter dadurch gekennzeichnet, daß die Klemmelemente (41 und 42) abstehende Hebelarme (45) besitzen, die mit dem Mittel (32) zum Einstellen des Anstellwinkels jedes Blattes verbunden sind, wodurch die geradlinige Bewegung des Mittels (32) in die Winklereinstellung des Blattanstellwinkels übertragen wird.

9. Drehflügel-Flugzeug nach einem der vorangehenden Ansprüche, weiter dadurch gekennzeichnet, daß das Mittel zum Einstellen der Anstellwinkel der Blätter Anstellwinkel-Steuerwellen (32) enthält, die jeweils einzeln innerhalb der jeweiligen Rotorwelle (33-36) angeordnet sind, daß jede Anstellwinkel-Steuerwelle (32) zur Übertragung der Anstellwinkel-Einstellbewegung zu dem zugeordneten Blatt geradlinig bewegbar ist und daß Antriebsmechanismen (53-56 oder 60) mit den Anstellwinkel-Steuerwellen (32) gekoppelt sind, um jeder Einstellwinkel-Steuerwelle differentielle kollektive Einstellwinkel-Einstellbewegung zu erteilen.



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