

PROCESS FOR REMOVING THERMAL BARRIER COATINGS

Background of the Invention

The present invention relates generally to a process for removing thermal barrier coatings from metal components and more particularly to a method for removing a thermal barrier ceramic coating from the cooling holes of a gas turbine engine component, such as a combustor chamber liner.

Gas turbine engines (aerospace and industrial) are designed such that their nickel and cobalt based superalloy components operate at temperatures very close to their melting points. Thermal barrier coatings (TBC) perform the important function of insulating components operating at elevated temperatures. Typical turbine components are combustion chamber (see combustion chamber 10 in FIGURE 1), ducts, discharge nozzles, turbine blades and nozzle guide vanes. TBCs are characterized by their very low thermal conductivity, the coating bearing a large temperature gradient when exposed to heat flow.

The most commonly applied TBC material is yttria stabilized zirconia (YSZ), which exhibits resistance to thermal shock and thermal fatigue up to 1150 degrees C. Typically the ceramic layer can be deposited by air plasma spraying (APS), low pressure plasma spraying (LPPS) or a physical vapor deposition (PVD) process, such as electron beam physical vapor deposition (EBPVD). It is common practice to pre-coat the substrate material with a bond coat. The bond coat accommodates residual stresses that might otherwise develop in the coating system, caused by the metallic substrate and the ceramic

TBC having different coefficients of thermal expansion, as well as providing oxidation and corrosion resistance. Typical bond coats include, but are not limited to, MCrAlY, wherein M is Ni, Co, Fe or mixtures thereof, or a diffusion aluminide or platinum aluminide coating.

The desire to increase the efficiency of gas turbine engines has led to an increase in the temperature in the combustion chamber and the hot section of the turbine engine. In order to compensate for the additional temperature, effusion hole cooling is often used where there is a significant heat load. *Effusion hole cooling of an engine component, such as a combustion chamber 10 as shown in FIGURE 1,* is accomplished by laser drilling small diameter (0.010 to 0.060 inch diameter) cooling holes 11 at specific angles and patterns that deliver the required cooling air to the engine component. Effusion hole cooling systems are typically used in conjunction with TBC coatings on engine components in order to achieve maximum benefit resulting from their ability to sustain high thermal gradients. Lowering the temperature of the metal substrate prolongs the life of the engine component. In addition, these cooling hole and TBC systems reduce the thermal gradients in the metal substrate thereby reducing the driving force for thermal fatigue. The benefit of these systems is realized in *greater component durability, higher gas temperature, performance and improved efficiency.*

Laser drilling (e.g. Nd: YAG laser) is used to drill and manufacture cooling holes in gas turbine engine hot section components. These parts are preferably protected by thermal barrier

coatings (TBC). The laser drilling process can manufacture the cooling holes by drilling through the component's metallic substrate and the TBC at the same time; however, laser induced damage occurs during this manufacturing process. Microstructural damage is generated at the TBC interface with the metallic bond coat and metallic substrate which results in TBC debonding and subsequent ceramic insulation coating loss (spallation) which is detrimental to the metallic substrate due to the high heat loads which adversely affects part durability and service life.

Various techniques have been developed to remove thermal barrier coatings from components during manufacture and repair, including air-cooled components. U.S. Patent No. 6,004,620, EP 1340587 A2 and U.S. Patent No. 6,620,457 B2 disclose a waterjet system with or without particulate media (abrasive or non-abrasive) utilizing a liquid-containing jet which operates at high fluid pressures ranging from 5000 pounds per square inch to 50,000 pounds per square inch in order to remove thermal barrier coating deposits. The waterjet process creates "minimal" wear and erosion of the underlying substrate after only a single cycle at 5000 pounds per square inch pressure. Additional cycles and/or increased pressures provides wear and erosion beyond what is considered minimal.

Summary of the Invention

Briefly, the present invention provides a process which uses an air jet containing non-abrasive particulate media at a low pressure which selectively removes thermal barrier coatings from components without damaging the metallic substrate. This process selectively removes thermal barrier coatings from the cooling holes of components.

Brief Description of the Drawings

Figure 1 represents a side view of the apparatus used to carry out the process of this invention.

Detailed Description Of The Invention

The present invention provides a process which utilizes a low pressure air-only jet with non-abrasive particulate media which is effective for the removal of thermal barrier coating deposits and does not result in wear erosion of the underlying metal substrate due to the low pressure process air and the inherent characteristics of the non-abrasive media that is used. The present process can be used an unlimited number of times on a component without significant wear and erosion.

This process utilizes a *non-abrasive particulate media*, preferably a dry spherical bead medium, under controlled low pressure air which effectively machines and removes thermal barrier coatings without adversely affecting the component metallic substrate. This process is effective in removing TBC coating that can be deposited in or on component features including, but not limited to, effusion air cooling holes and air cooling holes in general, as well as any component feature where TBC is not required or desired.

This process enables components to be manufactured or repaired by a laser drilling process (if required) which produces or restores air cooling holes prior to application or restoration of the thermal barrier coating. A primary beneficial attribute of this process is the ability to apply the thermal barrier coating after any required laser hole drilling thereby eliminating the laser induced damage which contributes to TBC loss (spallation) and subsequent thermal damage or distress to the component metallic substrate which adversely affects part durability and service life.

An additional benefit of this process is the surface enhancement of air cooling hole interiors and hole edges by smoothing and rounding any rough or sharp edges which significantly improves the airflow characteristics of the air cooling holes. Airflow testing of a combustion chamber wall has demonstrated a 14.6 percent increase of mass airflow after this process. Airflow test results of a combustion chamber liner wall after laser drilling and deburring showed a mass flow rate of 0.333093 LBM/S, (pounds per second mass flow) while airflow test results of the same wall after TBC application and use of this process showed a part mass flow rate of 0.382348 LBM/S.

The process is carried out using a dry air blast system 1 as shown in FIGURE 1 consisting of, but not limited to, an enclosure (not shown), rotary table 2, multiple air jet nozzles 3, a dry air supply pressure vessel and media processing unit 4, a mechanism for air jet nozzle movement 5, a media processing/recovery unit for filtration and spherical particle separation (not shown), and a programmable controller (not shown). The dry air jet system apparatus utilizes a media recovery unit which provides continuous media filtration and spherical particle separation during the process in order to maintain the efficiency of the process and produce the optimum surface enhancement in and adjacent to air cooling holes which is desirable in order to increase the discharge coefficient of the air cooling hole.

The process employs an air-only delivery system at a low pressure, preferably about 20 pounds per square inch (PSIG) to about 100 PSIG to flow a concentrated stream of non-abrasive particulate media to the work area in order to machine away and remove thermal

barrier coating deposits without adversely affecting the component's metallic substrate surface. The preferred non-abrasive media is a spherical media with at least 70%, preferably at least 95% the particles at a size in the range of about 0.002 to 0.010 inch (about 0.05 to 0.25 millimeters) diameter. Spherical glass media at a size of about 0.003 inch diameter can be used as the non-abrasive media.

Due to varying component geometry and design, the dry air jet system nozzle(s) may be articulated in any manner in order to achieve any position required to deliver the concentrated stream of spherical media to the work area in order to machine away and remove thermal barrier coating deposits from the component including but not limited to the interior, edges and surfaces of air cooling holes at varying angles of incidence. A manual dry air blast nozzle can also be utilized as required to perform the process. In removing the TBC from cooling holes the air jet can be directed at the cooling hole toward the surface opposing the surface having the thermal barrier coating, e.g. with a combustion chamber 10 shown in Figure 1 with a TBC coating on the exterior surface 12 the air jet would be directed at the cooling holes 11 from the interior surface. In addition to remove TBC from the cooling hole the air jet is aligned coaxially with the axis of the cooling hole (i.e. at substantially the same angle as the cooling hole).

The present invention can be applied to thermal barrier coated components during manufacture or repair. For the manufacture of new components, TBC can be applied after all other processing has been completed, including after laser drilling and machining. Since TBC application can be accomplished as one of the final manufacturing

processes utilizing the present invention, the quality and integrity of the TBC applied to the component is ensured and will possess the same quality level and characteristics as the representative test coupon that is normally processed with the component during TBC application. During component manufacture, this process can be used to remove TBC as required in order to accomplish any modification and/or repair which may be required to produce the component. For the repairs of service components, this process can be used to carefully remove TBC in order to prepare the part for inspection and repair. After repair processing is complete, including but not limited to welding, grinding, heat treatment and laser drilling and machining, the TBC can be reapplied and this process used as required to remove any unwanted TBC.

The air-only low pressure-media bead blast-hone process is a controlled system which uses very small diameter (in the 0.003 inch diameter range) spherical media at low air pressures in the range of about 20 PSIG to 100 PSIG for most TBC removal applications. Some TBC removal applications may require process parameter and/or media changes without departing from the scope of the claimed invention. The process is non-aggressive and does not cause wear or erosion to metallic substrates, is economical and produces consistent results. Airflow testing of cooling holes after application of this process has also demonstrated significantly improved component airflow characteristics.

Example

A jet engine hot section combustion chamber (manufactured from a cobalt or nickel based superalloy material) is manufactured having air cooling holes laser drilled therein. The gas path (hot side) surface of the component is first coated with a metallic bond coat which is predominantly nickel and containing chromium, aluminum and yttria (or another reactive element). The metallic bond coat is applied by plasma spraying to a thickness which is typically about 0.005 to 0.008 inch (about 0.13 to 0.020 mm). After bond coat application, the air cooling holes are manufactured utilizing a laser drilling process to create the desired angle of incidence in order to achieve the required air flow cooling characteristics for the component. After laser drilling, the component is cleaned to remove any laser slag or displaced material resulting from the laser drilling process. After laser drilling and cleaning, an optional bond coat layer of the same material can be applied by the plasma spray process to a thickness of about 0.001 to 0.003 inch. A ceramic top-coat layer which is predominantly zirconia partially stabilized with 6 to 8 percent of yttria is then applied by a plasma spray process to a thickness of about 0.005 to 0.020 inch (about 0.13 to 0.50 mm). During the application of the ceramic top-coat layer, the component's air cooling holes are partially blocked by the thermal barrier coating (ceramic top-coat layer) thereby restricting cooling air flow.

A Guyson Dry Air Blast System Model RSSA-8 configured as shown in Figure 1 is used to provide a dry air jet at a pressure of 40 to 60 PSIG utilizing spherical glass media in the 0.003 inch diameter range (at least 70 percent of the spherical particles possess a diameter of 0.003 inch). The dry air jet is directed to the metallic surface side (non-coated) of the component (opposing the

thermal barrier coated surface) at substantially the same angle of incidence as the air cooling hole to remove the thermal barrier coating deposits restricting cooling air flow.

The thermal barrier coating deposits are completely removed from the air cooling holes thereby providing the desired cooling air flow required for the component. In addition, the air cooling holes surfaces and edges are conditioned (smoothed and rounded) providing an approximate 15 percent increase in mass air flow by increasing the discharge coefficient of the air cooling holes.

What is claimed is:

1. A process for removing a thermal barrier ceramic coating from a metallic substrate surface of a component comprising:
directing an air jet at the thermal barrier coating on the substrate surface of the component, the jet containing a non-abrasive particulate media and being emitted from a nozzle at a low pressure insufficient to damage the substrate surface but sufficient to remove the thermal barrier ceramic coating.
2. Process of Claim 1 wherein the pressure of the air jet is from about 20 to 100 PSIG.
3. Process of Claim 2 wherein the media has a substantially spherical shape.
4. Process of Claim 3 wherein the spherical media particles have a diameter of from about 0.002 to 0.010 inches.
5. Process of Claim 4 wherein the media is glass beads.
6. Process of Claim 1 wherein the component is a turbine engine component.
7. Process of Claim 6 wherein the turbine engine component is a combustion chamber.
8. A process for removing a thermal barrier ceramic coating from a cooling hole of a metallic turbine engine component comprising:
directing an air jet at the cooling hole of the component, the jet containing non-abrasive particulate media and being emitted from a nozzle at a low pressure insufficient to

damage a metallic surface of the cooling hole but sufficient to remove the thermal barrier ceramic coating.

9. Process of Claim 8 wherein the pressure of the air jet is from about 20 to 100 PSIG.
10. Process of Claim 9 wherein the media has a substantially spherical shape.
11. Process of Claim 10 wherein the spherical media particles have a diameter of from about 0.002 to 0.010 inches.
12. Process of Claim 11 wherein the media is glass beads.
13. Process of Claim 12 wherein the turbine engine component is a combustion chamber.
14. Process of Claim 8 wherein the air jet is directed at the cooling hole toward a surface of the component opposing the surface having the thermal barrier coating.
15. Process of Claim 9 wherein the air jet is directed at the cooling hole at substantially the same angle as the cooling hole.
16. Process of Claim 8 wherein the air jet rounds the metallic edges of the cooling hole.
17. Process of Claim 8 wherein the cooling holes are drilled into the turbine component using a laser drilling process.
18. A process for forming cooling holes on a thermal barrier coated turbine engine component comprising:
drilling cooling holes into the component;
coating a surface of the component containing the cooling holes with a thermal barrier ceramic coating; and

directing an air jet at the cooling hole of the component, the jet containing non-abrasive particulate media and being emitted from a nozzle at a low pressure insufficient to damage a metallic surface of the cooling hole but sufficient to remove the thermal barrier coating.

19. Process of Claim 18 wherein the pressure of the air jet is from about 20 to 100 PSIG.
20. Process of Claim 19 wherein the media has a substantially spherical shape.
21. Process of Claim 20 wherein the spherical media particles have a diameter of from about 0.002 to 0.010 inches.
22. Process of Claim 21 wherein the media is glass beads.
23. Process of Claim 22 wherein the turbine engine component is a combustion chamber.
24. Process of Claim 16 wherein the air jet is directed at the cooling hole toward a surface of the component opposing the surface having the thermal barrier coating.
25. Process of Claim 18 wherein the air jet is directed at the cooling hole at substantially the same angle as the cooling hole.
26. Process of Claim 18 wherein the air jet rounds the metallic edges of the cooling hole.
27. Process of Claim 18 wherein the cooling holes are drilled through the turbine component using a laser drilling process.