CLAIMS

1. A vane for use in a gas turbine engine comprising:

an airfoil extending between a leading edge and a trailing edge, and being hollow;

a radially outer platform and a radially inner platform; with at least one of said radially inner and radially outer platforms having thicker portions circumferentially located beyond one of said leading edge and said trailing edge, and

thinner portions circumferentially beyond circumferential edges of said airfoil, and wherein the thicker portions are at a location where the airfoil merges into the radially inner platform.

2. The vane as set forth in claim 1, wherein said at least one platform is said radially inner platform.

3. The vane as set forth in claim 1, wherein said thicker portions vary in thickness, and extend circumferentially beyond circumferential edges of said airfoil.

4. The vane as set forth in claim 3, wherein a radial thickness of said thicker portion at a thickest portion compared to a radial thickness of thinner portions circumferentially spaced from said thicker portions is equal to or between about 1.1 and 5.0.

5. The vane as set forth in claim 1, wherein said thicker portions are provided by at least one thicker bump positioned axially beyond said leading edge, but circumferentially aligned with said airfoil.

6. The vane as set forth in claim 5, wherein there are at least a plurality of said bumps.

7. The vane as set forth in claim 6, wherein a radial thickness of a thickest portion of said bump is defined, and a ratio of said radial thickness of said thickest portion of said bump to a radially thickness of portions circumferentially spaced from said bump is equal to or between about 1.1 and 5.0.

8. The vane as set forth in claim 1, wherein said one of said leading edge and said trailing edge is said leading edge.

9. A mid-turbine frame comprising:

a plurality of vanes having airfoils extending between a radially inner and a radially outer platform;

said airfoil shape extending between a leading edge and a trailing edge, and being hollow; and

with at least one of said radially inner and radially outer platforms having thicker portions circumferentially located beyond said leading edge, and thinner portions circumferentially beyond circumferential edges of said airfoil, and wherein the thicker portions are at a location where the airfoil merges into the radially inner platform.

10. The mid-turbine frame as set forth in claim 9, wherein said thicker portions vary in thickness, and extend circumferentially beyond circumferential edges of said airfoil.

11. The mid-turbine frame as set forth in claim 10, wherein a radial thickness of said thicker portion at a thickest portion compared to a radial thickness of thinner portions circumferentially spaced from said thicker portions is equal to or between about 1.1 and 5.0.

12. The mid-turbine frame as set forth in claim 9, wherein said thicker portions are provided by at least one thicker bump positioned axially beyond said leading edge, but circumferentially aligned with said airfoil.

13. The mid-turbine frame vane as set forth in claim 12, wherein a radial thickness of a thickest portion of said bump is defined, and a ratio of said radial thickness of said thickest portion of said bump to a radially thickness of portions circumferentially spaced from said bump is equal to or between about 1.1 and 5.0.

14. The mid-turbine frame as set forth in claim 12, wherein there are at least a plurality of said bumps.

15. A gas turbine engine comprising:

at least a first upstream turbine rotor and a downstream turbine rotor, with a mid-turbine frame located intermediate said upstream and downstream turbine rotors;

the mid-turbine frame including a plurality of vanes having a radially outer platform and a radially inner platform, and an airfoil extending between a leading edge and a trailing

edge, and being hollow; and

at least one of said radially inner and radially outer platforms having thicker portions circumferentially located beyond said leading edge, and thinner portions circumferentially beyond circumferential edges of said airfoil.

16. The gas turbine engine as set forth in claim 15, wherein said at least one platform is said radially inner platform.

17. The gas turbine engine as set forth in claim 15, wherein said thicker portions vary in thickness, and extend circumferentially beyond circumferential edges of said airfoil.

18. The gas turbine engine as set forth in claim 17, wherein a radial thickness of said thicker portion at a thickest portion compared to a radial thickness of thinner portions circumferentially spaced from said thicker portions is equal to or between about 1.1 and 5.0.

19. The gas turbine engine as set forth in claim 15, wherein said thicker portions are provided by at least one thicker bump positioned axially beyond said leading edge, but circumferentially aligned with said airfoil.

20. The gas turbine engine as set forth in claim 19, wherein a radial thickness of a thickest portion of said bump is defined, and a ratio of said radial thickness of said thickest portion of said bump to a radially thickness of portions circumferentially spaced from said bump is equal to or between about 1.1 and 5.0.