CLAIMS

1. A component for a gas turbine engine, comprising:

a body portion;

a cooling circuit disposed within said body portion and including at least a first cavity, a miniature cooling cavity positioned within an outer skin of said body portion and in fluid communication with said first cavity, said miniature cooling cavity including an exit surface that extends through an exterior surface of said body portion;

at least one surface indicator visible near said exit surface;

wherein said at least one surface indicator is visible near at least one of a leading edge, a trailing edge, an inner diameter and an outer diameter of a discharge opening of said exit surface; and

wherein said at least one surface indicator is presented in a predetermined pattern that indicates whether the component is within an acceptable manufacturing tolerance.

2. The component as recited in claim 1, wherein said body portion is an airfoil of at least one of a blade and a vane.

3. The component as recited in claim 1, wherein said body portion is part of a blade outer air seal (BOAS).

4. The component as recited in claim 1, wherein said body portion is part of a mid-turbine frame.

5. The component as recited in claim 1, wherein said body portion is part of a combustor panel.

6. The component as recited in claim 1, wherein said exit surface includes at least one discharge opening.

7. The component as recited in claim 1, comprising a plurality of surface indicators visible near said exit surface.

8. The component as recited in claim 7, wherein a first portion of said plurality of surface indicators are located at a leading edge of a discharge opening of said exit surface and a second portion of said plurality of surface indicators are located at a trailing edge of said discharge opening.

9. The component as recited in claim 1, wherein said miniature cooling cavity is a microcircuit.

10. The component as recited in claim 1, wherein said miniature cooling cavity is a trailing edge, pressure side discharge core.