ABSTRACT

AIRCRAFT PROPULSION

A propulsion system 2 for an aircraft 1 comprises a gas turbine engine 10; a plurality of fuel tanks 50, 52 arranged to contain different fuels to be used to power the gas turbine engine 10, wherein the fuels have different calorific values; and a fuel manager 214. The fuel manager 214 is arranged to store information on the fuel contained in each fuel tank 50, 52 and to control fuel input to the gas turbine engine 10 in operation by selection of a specific fuel or fuel combination from one or more of the plurality of fuel tanks based on thrust demand of the gas turbine engine 10 such that a fuel with a lower calorific value is supplied to the gas turbine engine 10 at lower thrust demand.

To be reproduced with Figure 10 when published.