

MICRO-CIRCUIT PLATFORM

ABSTRACT OF THE DISCLOSURE

A gas turbine engine component, such as a high pressure turbine blade, has an airfoil portion, a platform, and micro-circuits within the platform for cooling at least one of a platform edge adjacent the pressure side of the airfoil portion and the trailing edge of the platform. The micro-circuits include a first micro-circuit on a suction side of the airfoil and a second micro-circuit on a pressure side of the airfoil. The micro-circuits within the platform achieve high thermal convective efficiency, high film coverage, and high cooling effectiveness.

Figure 2