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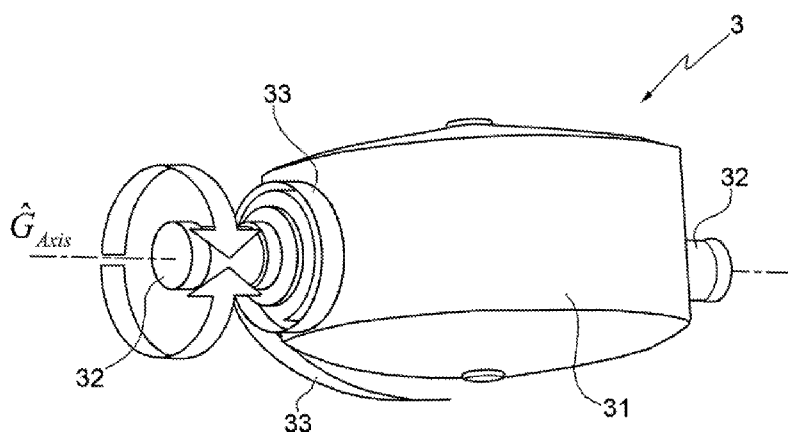


FIG.3

(57) Abstract: The present invention relates, in general, to attitude control and, in particular, to control of the attitude of a space platform, conveniently of a satellite and/or a spacecraft. In detail, an aspect of the present invention concerns the use, in an attitude control system (100), of several Control Moment Gyroscopes (3,41,42,43,44,61,62,63,64) with limited gimbal revolutions. Moreover, another aspect of the present invention concerns an improved logic for controlling a Control Moment Gyroscope assembly (4,6,120) of an attitude control system (100).