a first seal positioned along the endwall segment, wherein the first seal follows a shape of the first depression; and

a second seal positioned along the endwall segment opposite the first seal, wherein the second seal follows a shape of the second depression.

18. The assembly of claim 10, wherein the first and second depressions are spaced apart in the circumferential direction.

19. The assembly of claim 10, wherein at least one of the first and second depressions has a length that is approximately 93-94% of a root chord length of at least one of the plurality of airfoils.

~~19~~20. The assembly of claim 10, further comprising:

a protrusion formed along the flowpath surface, located in between the first and second depressions in a circumferential direction.

~~20. A rotor assembly for a gas turbine engine, the assembly comprising:~~

~~a rotor disk having one or more attachment features;~~

~~a plurality of airfoils extending from the rotor disk, each airfoil having a suction side and an opposite pressure side;~~

~~a spacer positioned between an adjacent pair of airfoils of the plurality of airfoils, the spacer comprising:~~

~~an endwall segment with a flowpath surface, wherein a perimeter of the flowpath surface includes a forward edge, an aft edge, a suction side edge and a pressure side edge, wherein the suction side edge is positioned adjacent to the suction side of one of the pair of airfoils separated by a suction side gap, and wherein the pressure side edge is positioned adjacent to the pressure side of the other of the pair of airfoils separated by a pressure side gap; and~~

~~a radial depression formed along the flowpath surface adjoining one of the suction side edge and the pressure side edge;~~

~~a first seal positioned at least partially along the suction side gap; and~~

~~a second seal positioned at least partially along the pressure side gap, wherein the first or second seal conforms to a shape of the depression.~~

~~21. The assembly of claim 20, wherein the spacer is mechanically attached to the rotor disk, and wherein the pressure side edge and the suction side edge of the endwall segment are each radiused.~~

~~22. The assembly of claim 20, further comprising:~~

~~a protrusion formed along the flowpath surface, located in between the first and second depressions in a circumferential direction.~~

~~23~~21. A spacer for a gas turbine engine, the spacer comprising:

an endwall segment having a flowpath surface; and

a stiffening structure located opposite the flowpath surface, the stiffening structure comprising:

a central rib; and

a plurality of additional ribs extending from the central ribs at acute angles.

~~24~~22. The spacer of claim ~~23~~21, wherein a perimeter of the flowpath surface includes a forward edge, an aft edge, a suction side edge and a pressure side edge, the spacer further comprising:

a first seal element positioned along the suction side edge; and

a second seal element positioned along the pressure side edge.