

Abstract:

A method for recontouring a compressor blade or a turbine blade (1) for a gas turbine, in which at least a portion of the leading edge (2) of the compressor blade or the turbine
5 blade (1) is systematically melted by a beam of energy (8) in such a way that the material hardens to form a new contour (5), essentially without the incorporation of additional material.

(Fig. 2)