METHOD FOR REPAIRING GAS TURBINE ROTOR BLADES

ABSTRACT OF THE DISCLOSURE

A method for replacing a portion (90) of a gas turbine engine rotor blade (50), the rotor blade having a contour defined by a blade first sidewall (70) and a blade second sidewall (70) includes cutting through the rotor blade such that a cut line (110) extends from a leading edge (74) of the blade to a trailing edge (76) of the blade, and between the first sidewall and the second sidewall, removing the portion of the rotor blade that is radially outward of the cut line, and coupling a replacement blade portion (120) to remaining blade portion (98) such that the newly formed rotor blade (150) is formed with a pre-determined aerodynamic contour.

FIG. 4