

COMPONENT MADE OF A FIBRE COMPOSITE MATERIAL COMPRISING
WOUND LAYERS

DESCRIPTION

The present invention relates to a component such as, for example, a crucible for melting a metal, made of a fibre composite material comprising at least three wound layers of a fibre material that are arranged one above the other, and a method for fabricating such a component.

Wound components made of fibre composite material are used in various fields of technology, for example, in the fabrication of melting crucibles or pressure containers. Conventionally, in order to fabricate such components, fibre strands optionally impregnated with binder are wound around a winding core and laid down on this. In this case, the fibre strands, which are also designated as rovings, can comprise different individual fibres made of carbon, glass, ceramic or aramid which can be embedded in a matrix of polymer or of ceramic, formed from binders, where the polymer matrix can subsequently be converted into a carbon matrix by carbonisation. Hence, the component formed from the wound layers, possibly after removal of the winding core and after appropriate curing and possibly carbonisation of the binder, can be constructed of carbon-fibre-reinforced plastic (CFP), carbon-fibre-reinforced carbon (CFC), carbon-fibre-reinforced ceramic or ceramic-fibre-reinforced ceramic (CMC).

Usually each wound layer of such a component comprises a plurality of windings which are arranged to run in the same direction, i.e. at least substantially parallel, adjacent to one another or overlapping one another so that through each wound layer at least one section of the winding core is completely covered by the fibre material, i.e. without the formation of recesses or intermediate spaces between

the windings. In this case, the individual wound layers are wound successively over the winding core, where the individual windings within each layer, as set out, run at least substantially parallel to one another. Here the winding angles of the windings of different wound layers can be the same or different. In the so-called cross-winding, the windings of adjacent wound layers are of the same magnitude but differ in their sign with respect to the same reference axis.

A component of this type or a complete winding comprising a plurality of wound layers arranged one above the other in a stack fashion and configured to each be fully covering is known, for example, from EP 0 333 013 B1.

A disadvantage of these known components is that no mechanical connection exists between the individual wound layers but between the individual wound layers there is at best a relatively weak interaction of chemical and/or physical binding forces, for example, in the form of a binder which adhesively bonds the different wound layers. For this reason, the stability, in particular mechanical strength such as, for example, the impact toughness or the interlaminar shear strength, of the known components of the type mentioned initially is in need of improvement.

It is therefore the object of the present invention to improve the stability and mechanical strength of wound fibre composite components.

According to the invention, this object is solved by a component having the features of claim 1.

According to the invention, a component made of a fibre composite material comprises at least three wound layers made of a fibre material that are arranged one over the

other, where each of the wound layers comprises one or more windings, wherein

- for each wound layer comprising more than one winding it holds that all windings of said wound layer have the same winding angle and are separated from each other at least in some sections to form intermediate spaces between the windings,
- all windings of at least two of the wound layers have the same first winding angle and
- all of the one or more windings of at least one wound layer, which is arranged between the at least two wound layers having the windings having a same first winding angle, have a second winding angle, where the magnitude of the second winding angle is different from the magnitude of the first winding angle, and where at least one of the windings having the second winding angle intersects with at least one of the windings having the first winding angle.

Wound layer in the sense of the present invention is/are understood as one or more windings which, when the wound layer comprises several windings, run at least substantially parallel to one another and therefore do not intersect one another. In the simplest case, therefore, a wound layer consists of a winding, where the windings of the wound layers adjacent to this wound layer necessarily have a different winding angle, i.e. do not run at least substantially parallel to the winding of the wound layer. Alternatively to this, a wound layer in the sense of the present invention can also comprise a plurality of windings running at least substantially parallel to one another, i.e. windings each having the same winding angle, where according to the invention at least in some sections, intermediate spaces are present between the individual

windings of one wound layer, i.e. the adjacently arranged fibre strand sections of one wound layer have a specific, preferably constant spacing from one another, in plan view of the component. Since all the windings of one wound layer have the same winding angle, the windings within a wound layer do not intersect. The windings within a wound layer preferably overlap apart from a central point about which the winding core is turned during winding. At the aforesaid central point, an unavoidable overlap of the deposited windings can occur.

A winding in this case designates the section of a wound layer which runs completely around the circumferential surface of the winding core or the circumferential surface of the component once. In other words, a winding corresponds to a section of the fibre material deposited by a double stroke of the winding machine.

Winding angle is understood to be that angle at which the individual windings are deposited relative to an in-principle arbitrary reference axis on the winding core. In other words, the winding angle is the angle which the longitudinal axis of the winding occupies in relation to the reference axis at the instant of depositing on the winding core even if the orientation of the winding can still vary slightly due to some slippage of the fibre strand after depositing. If the winding core is configured to be cylindrical, no slippage of the fibre strand occurs so that in this case the winding angle is the angle which the longitudinal axis of the deposited winding adopts in relation to the reference axis.

With reference to the finished component, this means that for components having at least one cylindrically symmetrical section, where the cylindrically symmetrical section or sections account for at least 20% of the total surface of the component, the winding angle in the sense of

the present invention is the angle which the longitudinal axis of the section of the winding deposited on the cylindrically symmetrical section(s) adopts in relation to the reference axis, regardless of the orientation of the winding in the sections of the component different from the cylindrically symmetrical sections. For all other components, the term winding angle in the sense of the present invention means the angle between the winding and the geodetic line which completely spans the circumferential surface of the winding core or the circumferential surface of the component in the case of a rotationally symmetrical winding core or component parallel to the axis of rotation of the winding core or component and in the case of a non-rotationally symmetrical winding core or component parallel to the longitudinal axis of the winding core or component. From this it follows, taking into account the preceding definition of the term winding angle, that all windings run substantially parallel whereby it is understood in the sense of the present invention that the standard deviation of the winding angle present at any point of the winding divided by the arithmetic mean of all corresponding winding angles is less than 0.5. This permissible standard deviation from an ideally rectilinear configuration takes into account the fact that during the deposition at a defined angle in particular on regions of curvature such as the transition region between a cylindrical region and a conical region of the circumferential surface of the winding core or component, the windings can slip minimally with the result that a certain slight change in the winding angle can occur inside the winding. In addition to the preceding standard deviation, it also applies according to the present invention that the difference between the maximum winding angle and the minimal winding angle of each winding is less than 1° , preferably at most 0.8° , particularly preferably at most 0.5° , and quite particularly preferably at most 0.2° .

As set out hereinbefore, the winding angle of all the windings of one wound layer is the same amongst one another. In order to again allow for a possible slippage of sections of individual windings after depositing, in the sense of the present invention it is understood by a same winding angle that the winding angles of all the windings inside a wound layer relative to a geodetic reference line which completely spans the circumferential surface of the winding core or the circumferential surface of the component in the case of a rotationally symmetrical winding core or component parallel to the axis of rotation of the winding core or component and in the case of a non-rotationally symmetrical winding core or component parallel to the longitudinal axis of the winding core or component, differ from one another by less than 1° , preferably by at most 0.8° , particularly preferably by at most 0.5° and quite particularly preferably by at most 0.2° .

In a component according to the invention, there is therefore between at least two wound layers having a same first winding angle a wound layer having a second different winding angle. As a result of the intermediate spaces provided between the individual windings of one wound layer according to the invention, the winding patterns of these wound layers set by the different winding angles overlap in a crossover fashion. Since the magnitudes of the first and the second winding angle differ, at the transition of the corresponding wound layers, not only the sign of the winding angle is changed as is the case in the known cross windings. According to the invention, intersections of the windings distributed uniformly over the wound region therefore do not exist, as in a cross winding; on the contrary, in the component according to the invention, non-uniformly distributed intersections of the individual wound layers exist. As a result, among other things a better load distribution and therefore a higher mechanical strength of

the component is achieved. This increased mechanical strength already exists when a single fully-covering arrangement of windings is provided.

Due to the type of arrangement of the individual wound layers according to the invention, the sharp separation of the individual wound layers such as occurs in the windings known from the prior art is eliminated so that in each case the fibres of one wound layer engage in one or more of the wound layers located thereabove and/or therebelow. As a result, to a certain extent mixed layers or combination layers are formed where as a result, a single continuous total winding is achieved. Thus, a fabric-like braided structure is obtained in which the interlaminar binding is improved compared with the corresponding structures known from the prior art.

Whereas in wound fibre composite components according to the prior art, wound layers significantly separated from one another therefore exist, which are at best bound to one another via chemical and/or physical interactions, in the component according to the invention there is a mechanical interlocking or interleaving between the individual wound layers, whereby a substantially higher stability and mechanical strength, in particular an improved impact toughness and damage tolerance, of the component is achieved. A component according to the invention can therefore also be used for application having maximum loading requirements such as, for example, as a bullet-proof panel.

The side of the intermediate spaces present between the individual windings, i.e. the spacing of the individual fibre strands of a wound layer in plan view of the wound layer as well as the sequence of the individual winding angles or winding patterns between the individual winding

layers can fundamentally be varied widely depending on the requirements of the application.

Preferably the complete winding formed from all the wound layers arranged one above the other is free from intermediate spaces in plan view. In other words, the wound layers arranged one above the other completely cover the wound section of the winding core or the wound circumferential surface. The wound layers thus form a recess-free component wall, for example as vessel wall for a crucible. The intermediate spaces between the windings of a wound layer therefore do not serve to produce a component having a perforated wall but for the interleaving receipt of windings of other wound layers.

According to one embodiment of the invention, in each wound layer which comprises more than one winding, the ratio of the sum of the areas of the intermediate spaces between the windings in the wound layer to the total area of the wound layer is 0.05000 to 0.99999, preferably 0.90000 to 0.99999, particularly preferably 0.99000 to 0.99999 and quite particularly preferably is 0.99990 to 0.99999. Such area ratios have proved particularly favourable with a view to the stability and mechanical strength of the component.

According to the invention, the winding angles of all the windings within a wound layer differ from one another by less than 1° , preferably by at most 0.8° , particularly preferably by at most 0.5° and quite particularly preferably by at most 0.2° . This favours a uniform mechanical strength of the component.

On the other hand, the magnitude of the second winding angle differs from the magnitude of the first winding angle according to the invention by at least 1° , preferably by at least 2° , particularly preferably by at least 3° , quite particularly preferably by at least 4° and most preferably

by at least 5°. In order to achieve a defined interleaving between the wound layers, it is specifically favourable to avoid arrangements of wound layers having all too similar winding patterns.

Preferably at least one winding of at least one of the wound layers at least in some sections engages in at least one intermediate space of a wound layer arranged below and/or above the wound layer. The windings of the corresponding wound layers therefore do not lie strictly stacked upon one another but sections of windings of one wound layer each penetrate into another wound layer so that a firm mechanical connection between the wound layers is obtained.

According to a particularly preferred embodiment of the invention, all the windings of at least one of the wound layers having the windings having the first winding angle engage in the intermediate spaces of the at least one wound layer having the at least one winding having the second winding angle and/or in the intermediate spaces of the further wound layer having the windings having the first winding angle. For this purpose, preferably the windings of the individual wound layers having the first winding angle are arranged offset to one another so that the windings of one of the wound layers containing windings having the first winding angle are arranged over the intermediate spaces of another of the wound layers containing windings having the first winding angle. As a result, a uniform intermeshing of windings relative to the circumferential surface of the winding core or the component and therefore a particularly high strength of the component is achieved.

Each of the wound layers can comprises 1 to 2,000, preferably 1 to 250, particularly preferably 1 to 150 and quite particularly preferably 1 to 10 windings. The number

of windings is determined in detail by the requirements of the particular application.

Furthermore the component preferably comprises at least 10, preferably 200 to 10,000, particularly preferably 500 to 10,000 and quite particularly preferably 2,000 to 5,000 wound layers arranged one above the other. By adapting the number of wound layers arranged one above the other, a desired wall thickness or a desired mechanical strength of the component can be achieved with predefined strength of the fibre strands.

According to a further preferred embodiment of the invention, in each case one wound layer having the windings having the first winding angle and one wound layer having the windings having the second winding angle are arranged directly above one another in alternating sequence and specifically particularly preferably directly above one another. In this embodiment the resulting winding pattern resembles a cross winding where, however in contrast to a cross winding, the desired interleaving effect is achieved by the different winding angles in terms of magnitude.

At least one additional wound layer can be arranged between the at least two wound layers having the windings having the first winding angle apart from the wound layer having the windings having the second winding angle. By providing such "intermediate layers", the mechanical properties of the component can be influenced in a desired manner. The at least one additional wound layer can comprise one or more windings wherein all these windings have a third winding angle whose magnitude differs from the magnitude of the first winding angle and from the magnitude of the second winding angle. In this way, three or more different winding angles can be combined with one another in order to further increase the interleaving effect between the wound layers and thus the component strength.

Preferably all the windings having the second winding angle intersect with at least one winding having the first winding angle. The windings having the second winding angle are therefore not arranged spatially remote from the windings having the first winding angle on the winding core. On the contrary, the windings having different winding angles are superposed on one and the same section of the winding core or component, where as a result of the different winding angles cross-overs inevitably result which ultimately improve the connection between the wound layers.

Preferably the fibres are selected from the group which consists of carbon fibres, ceramic fibres, glass fibres, polymer fibres and mixture of two or more of the aforesaid materials, wherein the polymer fibres are preferably polyamide fibres, in particular aramid fibres, polyester fibres, polypropylene fibres, polyacrylonitrile fibres, fibres of oxidised polyacrylonitrile and fibres of copolymers of two or more of the aforesaid materials and mixtures of two or more of the aforesaid materials. Such fibre materials are suitable for providing wound components having high mechanical strength. The mean diameter of the fibres in this case can be, for example, 3 to 20 μm . Fibre bundles can also be used where the average number of fibres per deposited fibre bundle can be between 500 and 400,000, preferably between 1,000 and 50,000 and particularly preferably between 3,000 and 50,000.

A component according to the invention can in particular consist of a carbon-fibre-reinforced carbon (CFC), a carbon-fibre-reinforced plastic (CFP), a carbon-fibre-reinforced ceramic or a ceramic-fibre-reinforced ceramic (CMC). Such composite materials are characterised among other things by a high mechanical and thermal stability, a

high shock resistance, a high chemical resistance and a high chemical purity.

The wound layers arranged one above the other form an unsupported winding. That is, the winding core used for winding the fibre material is removed before completing the fabrication of the component so that a cavity enveloped by the wound layers is formed. In this way, the component can be used as a hollow profile or as a container.

According to a further specific embodiment, the component is configured as a crucible for the melting and crystallisation of a metal and/or semi-metal.

The invention also relates to a method for fabricating a component made of a fibre composite material, in which at least three wound layers of a fibre material are wound on a winding core, wherein:

- for each of the wound layers one or more windings are wound, wherein all the windings of one wound layer which has more than one winding, are wound with the same winding angle and are separated from each other onto the winding core at least in some sections to form intermediate spaces between the windings,
- the windings of at least two of the wound layers having the same first winding angle are wound on the winding core and
- all of the one or more windings of at least one wound layer having a second winding angle, whose magnitude differs from the magnitude of the first winding angle are wound on the winding core in such a manner that this wound layer is arranged between the at least two wound layers having the windings having the first winding angle and at least one of the windings having

the second winding angle intersects with at least one of the windings having the first winding angle.

The superposition of different winding patterns in conjunction with the intermediate spaces left inside the wound layers enables an intermeshing of the individual wound layers and thus a certain interleaving or interlocking effect which increases the mechanical stability of the fabricated component.

Preferably the windings of the at least three wound layers are wound on the winding core in such a manner that they completely cover at least a section of the winding core. This section of the winding core forms a continuous recess-free wall on the finished component.

For each wound layer the windings of the fibre material are wound on the winding core separated from one another at least in some sections to form intermediate spaces between the windings so that the ratio of the sum of the areas of the intermediate spaces between the windings in the wound layer to the total area of the wound layer is 0.05000 to 0.99999, preferably 0.90000 to 0.99999, particularly preferably 0.99000 to 0.99999 and quite particularly preferably is 0.99990 to 0.99999. Such surface ratios have provided particularly favourable in regard to the stability of the component.

A method according to the invention can comprise as further steps:

- impregnating the fibre material before, during or after the winding with binder and
- curing the binder after the impregnating and after the winding of the fibre material.

The impregnation of the fibre material preferably takes place during the winding. Alternatively to this, the winding can also take place with already-pre-impregnated and/or stabilised fibre material (prepegs). Fundamentally, all binders known to the person skilled in the art can be used, where good results are obtained in particular if the matrix material is selected from the group which consists of epoxide resins, phenol resins, vinylester resins, polyester resins, polyurethane resins, benzoxazine resins, novolacs, cyanate ester resins, bismaleimide resins, bisoxalones, polyolefins such as, for example, polypropylene, technical thermoplastics such as, for example, polyamides and any mixtures of two or more of the aforesaid materials. Furthermore, the binders can comprise suitable fillers such as, for example, carbon or graphite particles or ground or short-cut fibres. In addition, a suitable additive such as, for example, hardeners, accelerators, emulsifiers, internal separating agents and the like can be added to the binder.

The curing can take place thermally and/or chemically depending on the type of binder.

After the curing, a carbonisation and/or graphitisation can be carried out in order to convert the matrix material into carbon. In this way, carbon-fibre-reinforced carbon composite materials can be produced. In this case, the carbonisation is preferably carried out at a temperature of 400 to 1,200°C and particularly preferably of 600 to 1,000°C, whereas the graphitisation is preferably carried out at a temperature of 1,400 to 3,000°C and particularly preferably of 1,600 to 2,400°C.

In order to further increase the carbon content of the matrix or increase the density of the fibre composite materials, after the carbonisation or graphitisation another one or more liquid compaction steps can be carried

out using suitable resins and/or pitches. After each such liquid compaction step, a carbonisation and/or graphitisation is carried out. A method according to the CVI principle ("chemical vapour infiltration") can also be used in order to further increase the carbon fraction in the matrix.

According to a specific embodiment of the invention, for the winding endless fibre strands containing carbon fibres are laid on the winding core.

Before the winding of the wound layers on the winding core in particular a program can be created and/or configured which contains control instructions relating to the winding for an automatic winding machine, wherein the winding of the windings is accomplished by executing the control instructions. The winding process per se can then be carried out fully automatically.

The invention is explained further hereinafter with reference to examples which explain this but do not restrict it by reference to the drawings.

Fig. 1 is a side view of a winding core on which a first wound layer of a fibre material according to a first exemplary embodiment is wound.

Fig. 2 shows the winding core according to Fig. 1 with an additional second wound layer wound over the first wound layer.

Fig. 3 shows the winding core according to Fig. 1 with an additional third wound layer wound over the second wound layer.

Fig. 4 shows the winding core according to Fig. 1 with an additional fourth wound layer wound over the third wound layer.

Fig. 5 is a side view of a winding core on which two wound layers of a fibre material arranged one above the other are wound according to a second exemplary embodiment.

Fig. 6 is a perspective view of the winding core according to Fig. 5.

Fig. 7 is a further perspective view of the winding core according to Fig. 5.

Fig. 8 is a side view of a section of a winding pattern deposited on a winding core according to another exemplary embodiment.

Figs. 9a to 9d show side views of a section of a winding core on which two wound layers of a fibre material (Fig. 9a), four wound layers of a fibre material (Fig. 9b), six wound layers of a fibre material (Fig. 9c) and eight wound layers of a fibre material (Fig. 9d) according to another exemplary embodiment are wound.

According to a first exemplary embodiment of the present invention shown in Figs. 1 to 4, a crucible for melting a metal is fabricated from a fibre composite material by winding a plurality of wound layers of a fibre material onto a winding core or winding mandrel 11. The winding core 11, shown only in part, is a metal hollow body having a cylindrical jacket section 13 and a dome-shaped cover section 15. The winding core 11 is mounted rotatably about an axis of rotation R on a supporting frame not shown,

where a feed head of a winding machine also not shown is disposed such that it can deposit endless fibre strands 17 composed of binder-impregnated carbon fibres on the winding core 11. in this way windings 23 which each run once completely around the winding core 11 are formed. During the deposition the endless fibre strands are impregnated with a curable polymer-based binder.

By moving the feed head and by turning the winding core 11 about the axis of rotation R during the deposition of the impregnated endless fibre strands 17 a certain winding angle β is set between a winding 23 and the geodetic line which completely spans the circumferential surface of the winding core 11 parallel to the axis of rotation R.

In this way a first wound layer 20a is wound, whose windings 23 all have a winding angle β of 45° . The endless fibre strands 17 are deposited at a distance from one another so that intermediate spaces 25 exist between the individual windings 23.

In order to achieve a predetermined winding angle which would not be accessible as a result of a relatively small slipping friction of the wound layer on the winding core or on the preceding wound layers, pins or a pin rim can additionally be provided on the winding core, which hold or stabilize the wound layer in a suitable position in relation to the winding core. The use of pins is also particularly advantageous if the change in the winding angle from one wound layer to a next is so large that at as a result of the relatively small slipping friction, no stable holding of the wound layers on the winding core or the preceding wound layers would be possible.

Then, according to Fig. 2 a second wound layer 20b is wound over the first wound layer 20a onto the winding core 11,

where the windings 23 of this second wound layer 20b have a winding angle of 0° .

Then according to Fig. 3, a third wound layer 20a is wound over the second wound layer 20b onto the winding core 11. The windings 23 of the third wound layer 20c like the windings 23 of the first wound layer 20a have a winding angle of 45° but are relatively offset to the windings 23 of the first wound layer 20a so that they are located above the intermediate spaces 25 of the first wound layer 20a. In so doing, the windings 23 of the second wound layer 20b are pressed into the intermediate spaces 25 to a certain degree.

Furthermore, according to Fig. 4 a fourth wound layer 20d is wound over the third wound layer 20c onto the winding core 11. The windings 23 of the fourth wound layer 20d like the windings 23 of the second wound layer 20b have a winding angle of 0° but are offset relative to the windings 23 of the second wound layer 20b so that they are located above the intermediate spaces 25 of the second wound layer 20b. The windings 23 of the third wound layer 20b are thereby again pressed into corresponding intermediate spaces 25 located thereunder.

As a result of the two different winding angles, the windings 23 of the first wound layer 20a intersect with the windings 23 of the second wound layer 20b and the fourth wound layer 20d. Likewise, the windings 23 of the third wound layer 20c intersect with the windings 23 of the second wound layer 20b and the fourth wound layer 20d.

The two alternatively winding angles 0° and 45° are merely shown as an example. An alternative exemplary embodiment of the invention provides that the windings 23 of the first, second, third and fourth wound layer 20a, 20b, 20c, 20d have winding angles which are 0.3° , 2° , 5° and 6° .

Further wound layers are then wound over the existing wound layers 20a, 20b, 20c, 20d until the windings 23 of all wound layers arranged one above the other completely cover the jacket section 13 and the cover section 15 of the winding core 11.

Finally, the binder of the impregnated endless fibre strands 17 is thermally or chemically cured and the winding core 11 is removed. This can be accomplished for example by separating a section of the winding formed opposite the cover section 15 and removing the winding core 11 from the winding so that a hollow body of wound layers arranged one above the other is formed, which forms a crucible. Before or after removing the winding core 11, a carbonisation and/or graphitisation can additionally take place in order to convert the binder into amorphous carbon or graphite.

Figures 5 to 7 show another winding core 11 with deposited binder-impregnated endless fibre strands 17, where it can be seen here that the windings are guided over a front-side opening 33 of the winding core 11. A rotatable shaft 29 which defines the axis of rotation R extends through the opening 33. It is additionally deduced from Fig. 6 that unlike in the exemplary embodiment shown in Figs. 1 to 4, all the windings 23 intersect at a central point 31 at which the axis of rotation R passes through the cover section 15 of the winding core 11.

For the automatic execution of the winding, before the winding of the windings 23 on the winding core 11 a computer program is created and/or configured which contains control instructions relating to the winding for the drives of the rotatable shaft 29 and the feed head. The previously described winding of the windings 23 can be carried out fully automatically by executing the control instructions.

Figure 8 shows a side view of a section of a winding pattern deposited on a winding core according to another exemplary embodiment. It can be seen from this figure how the individual wound layers 20a, 20b of windings 23 intersect with quantitatively different winding angles in each case, where the intersections are distributed non-uniformly with the result that the desired interleaving effect according to the invention is achieved as a result of which the mechanical stability and mechanical strength, in particular impact toughness and damage tolerance of the component are increased.

Figures 9a to 9d show side views of a section of a winding core where two wound layers 20a, 20b are deposited in the winding core section shown in Fig. 9a, four wound layers 20a, 20b, 20c, 20d are deposited on the winding core section shown in Fig. 9b, six wound layers 20a, 20b, 20c, 20d, 20e, 20f are deposited on the winding core section shown in Fig. 9c and eight wound layers 20a, 20b, 20c, 20d, 20e, 20f, 20g, 20h are deposited on the winding core section shown in Fig. 9d. Each two of these wound layers 20a and 20b or 20c and 20d or 20e and 20f or 20g and 20h have the same winding angle in terms of magnitude but differ in regard to the sign whereas the other wound layers each have winding angles which differ from this in terms of magnitude. These cross-wound layers in the sense of the present invention each comprise two different wound layers. With this winding pattern non-uniformly distributed intersections are also obtained with the result that the desired interleaving effect according to the invention is achieved as a result of which the mechanical stability and mechanical strength, in particular impact toughness and damage tolerance, of the component are increased.

REFERENCE LIST

11	Winding core
13	Jacket section
15	Cover section
17	Endless fibre strand
20a, 20b, 20c, 20d	Wound layer
20e, 20f, 20g, 20h	Wound layer
23	Winding
25	Intermediate space
27	Intersection point
29	Rotatable shaft
31	Central point
33	Front-side opening
R	Axis of rotation
β, γ, δ	Winding angle

CLAIMS

1. A component comprising a fibre composite material and having at least three wound layers (20a, 20b, 20c, 20d) comprising a fibre material and arranged one above another, characterised in that:
 - the wound layers (20a, 20b, 20c, 20d) have one or more windings (23), wherein in each wound layer (20a, 20b, 20c, 20d) having more than one winding (23), all the windings (23) of said wound layer (20a, 20b, 20c, 20d) have the same winding angle and are separated from each other in at least one portion, whereby gaps (25) are formed between the windings,
 - all the windings of at least two of the wound layers (20a, 20c) have an identical first winding angle (β), and
 - all of the one or more windings (23) of at least one wound layer (20b), which is arranged between the at least two wound layers (20a, 20c) having the windings (23) having the same first winding angle (β), have a second winding angle (γ), wherein the magnitude of the second winding angle (γ) is different from the magnitude of the first winding angle (β), and wherein at least one of the windings (23) having the second winding angle (γ) intersects with at least one of the windings (23) having the first winding angle (β), and at least one winding (23) of at least one of the wound layers (20a, 20b, 20c, 20d) engages, in at least one portion, in at least one gap (25) of a wound layer (20a, 20b, 20c, 20d) arranged above and/or below that wound layer.

2. The component according to claim 1, characterised in that the overall wound body formed from all the wound layers (20a, 20b, 20c, 20d) arranged one above another has no gaps when seen from above.
3. The component according to claim 1 or 2, characterised in that in each wound layer (20a, 20b, 20c, 20d) having more than one winding (23), the ratio of the sum of the areas of the gaps (25) between the windings (23) in the wound layer (20a, 20b, 20c, 20d) to the total area of the wound layer (20a, 20b, 20c, 20d) is from 0.05000 to 0.99999.
4. The component according to at least one of claims 1 to 3, characterised in that the winding angles (β) of all the windings (23) within a single wound layer (20a, 20b, 20c, 20d) differ from one another by less than 1° .
5. The component according to at least one of claims 1 to 4, characterised in that the magnitude of the second winding angle (γ) differs from the magnitude of the first winding angle (β) by at least 1° .
6. The component according to at least one of claims 1 to 5, characterised in that all the windings (23) of at least one of the wound layers (20a) having windings having the first winding angle (β) engage in the gaps in a wound layer (20b) having at least one winding (23) having the second winding angle (γ), and/or in the gaps in a further wound layer (20c) having windings (23) having the first winding angle (β).
7. The component according to at least one of claims 1 to 6, characterised in that each of the wound layers

(20a, 20b, 20c, 20d) has from 1 to 2,000 windings (23).

8. The component according to at least one of claims 1 to 7, characterised in that the component has up to at least 10 wound layers (20a, 20b, 20c, 20d) arranged one above another.
9. The component according to at least one of claims 1 to 8, characterised in that each wound layer (20a, 20c) having the windings (23) having the first winding angle (β), and wound layers (20c, 20d) having the windings (23) having the second winding angle (γ), are arranged alternately and directly above one another.
10. The component according to at least one of claims 1 to 8, characterised in that at least one additional wound layer is arranged between the at least two wound layers (20a, 20c) having the windings (23) having the first winding angle (β), in addition to the wound layer (20b) having the windings (23) having the second winding angle (γ).
11. The component according to claim 10, characterised in that the at least one additional wound layer has one or more windings (23), wherein all these windings have a third winding angle (δ) whose magnitude differs from the magnitude of the first winding angle (β) and from the magnitude of the second winding angle (γ).
12. The component according to at least one of claims 1 to 11, characterised in that all the windings (23) having the second winding angle (γ) intersect with at least one winding (23) having the first winding angle (β).

13. The component according to at least one of claims 1 to 12, characterised in that the fibres are selected from the group comprising carbon fibres, ceramic fibres, glass fibres, polymer fibres and mixtures of two or more of the aforesaid materials, wherein the polymer fibres are polyamide fibres, aramid fibres, polyester fibres, polypropylene fibres, polyacrylonitrile fibres, fibres comprising oxidised polyacrylonitrile, fibres comprising copolymers of two or more of the aforesaid materials, and fibres comprising mixtures of two or more of the aforesaid materials.
14. The component according to at least one of claims 1 to 13, characterised in that the fibre composite material is a carbon-fibre-reinforced carbon (CFC), a carbon-fibre-reinforced synthetic resin (CFK), a carbon-fibre-reinforced ceramic or a ceramic-fibre-reinforced ceramic (CMC).
15. The component according to at least one of claims 1 to 14, characterised in that the wound layers (20a, 20b, 20c, 20d) arranged one above another form an unsupported wound body.
16. The component according to at least one of claims 1 to 15, characterised in that the component is formed as a crucible for the melting and crystallisation of a metal and/or a semi-metal.
17. A method for fabricating a component comprising a fibre composite material, in which fabrication method at least three wound layers (20a, 20b, 20c, 20d) comprising a fibre material are wound on a winding core (11), said method of fabricating a component being characterised in that:

- in each of the at least three wound layers (20a, 20b, 20c, 20d) one or more windings (23) are wound, wherein all the windings (23) of a wound layer (20a, 20b, 20c, 20d) which has more than one winding (23) are wound onto the winding core (11) with the same winding angle (β) and separated from each other in at least one portion to form gaps (25) between the windings (23),
- the windings (23) of at least two wound layers (20a, 20c) are wound on the winding core (11) with a same first winding angle (β), and
- all of the one or more windings (23) of at least one wound layer (20b) that have a second winding angle (γ) whose magnitude differs from the magnitude of the first winding angle (β), are wound on the winding core (11) in such a manner that this wound layer (20b) is arranged between the at least two wound layers (20a, 20c) having windings (23) having the first winding angle (β), and in such manner that at least one of the windings (23) having the second winding angle (γ) intersects with at least one of the windings (23) having the first winding angle (β); and so that in this case, at least one winding (23) of at least one winding layer (20a, 20b, 20c, 20d) engages, in at least one portion, in at least one gap (25) of a winding layer (20a, 20b, 20c, 20d) arranged above and/or below that winding layer.

18. The method of fabricating a component according to claim 17, characterised in that the windings (23) of the at least three wound layers (20a, 20b, 20c, 20d) are wound on the winding core (11) in such a manner

that they completely cover at least a portion of the winding core (11).

19. The method of fabricating a component according to claim 17 or 18, characterised in that each wound layer (20a, 20b, 20c, 20d) is wound on the winding core (11) in such manner that the ratio of the sum of the areas of the gaps between the windings (23) in a wound layer (20a, 20b, 20c, 20d) to the total area of the wound layer (20a, 20b, 20c, 20d) is from 0.05000 to 0.99999; and in such manner that at least two windings (23) of the fibre material are separated from one another in at least one portion, whereby a gap (25) is formed between the windings (23).
20. The method of fabricating a component according to at least one of claims 17 to 19, characterised in having further processing steps of:
 - impregnating the fibre material with a binder before, during or after the winding process, and
 - curing the binder after the impregnating and the winding of the fibre material.
21. The method of fabricating a component according to claim 20, characterised in that the curing is carried out thermally and/or chemically.
22. The method of fabricating a component according to at least one of claims 17 to 21, characterised in that for the winding process, endless fibre strands (17) containing carbon fibres are laid on the winding core (11).

23. The method of fabricating a component according to at least one of claims 17 to 22, characterised in that a program is created or configured before winding the wound layers (20a, 20b, 20c, 20d) on the winding core (11), this program containing control commands for an automatic winding machine and that relate to the winding process, wherein the winding of the windings (23) is accomplished by executing the control commands.